

priority project in each state, up to 7 points for the second highest priority project in each state and up to 5 points for the third highest priority project that address unforeseen exigencies or emergencies, such as the loss of a community facility due to an accident or natural disaster, or other areas of need in their particular state. The State Director will place written documentation in the project file each time the State Director assigns these points—Up to 10 points.

(i) *Administrator discretionary points.* The Administrator may award up to 20 discretionary points for projects to address geographic distribution of funds, emergency conditions caused by economic problems, natural disasters and other initiatives identified by the Secretary—Up to 20 points.

#### **§ 3570.274 Fund disbursement.**

The Agency will make payments under this agreement in accordance with 2 CFR 200.305. All requests for advances or reimbursements must be in compliance with 2 CFR 200.306 and include any required matching fund usage.

#### **§ 3570.275 Grant cancellation or major changes.**

Any change in the scope of the Project, budget adjustments of more than 10 percent of the total budget, and any other significant change in the Project must be in compliance with 2 CFR 200.308 and 200.339. The changes must be requested in writing and approved by the Agency in writing. Any change not approved may be cause for termination of the grant.

#### **§ 3570.276 Reporting.**

(a) The Grantee must provide periodic reports as required by the Agency. A financial status report, SF 425 “Federal Financial Report,” and a project performance report will be required as provided in the grant agreement. The financial status report must show how grant funds and matching funds have been used to date. A final report may serve as the last report. Grantees shall constantly monitor performance to ensure that time schedules are being met and projected goals by time periods are being accomplished. The Project performance reports shall include, but are not limited to, the following:

- (1) A description of the activities that the funds reflected in the financial status report were used for;
- (2) A comparison of actual accomplishments to the objectives for that period;
- (3) Reasons why established objectives were not met, if applicable;

(4) Problems, delays, or adverse conditions which will affect attainment of overall program objectives, prevent meeting time schedules or objectives, or preclude the attainment of particular objectives during established time periods. This disclosure shall be accomplished by a Statement of the action taken or planned to resolve the situation;

(5) Objectives and timetables established for the next reporting period;

(6) A summary of the race, sex, and national origin of the Ultimate Recipients;

(7) The final report will also address the following:

(i) What have been the most challenging or unexpected aspects of this grant?

(ii) What advice would you give to other organizations planning a similar grant? What are the strengths and limitations of this grant? If you had the opportunity, what would you have done differently?

(iii) Are there any post-grant plans for this Project? If yes, how will they be financed?

(b) [Reserved]

#### **§ 3570.277 Audit or financial statement.**

The Grantee will provide an Audit report or financial Statement in accordance with 2 CFR 200.500–200.517 and as follows:

(a) Grantees expending \$750,000 or more Federal funds per fiscal year will submit an Audit conducted in accordance with 2 CFR parts 200, 215, 220, 225, 230 and 400, “Uniform Administrative Requirements, Cost Principles, and Audit Requirements for Federal Awards.”

(b) Grantees expending less than \$750,000 will provide annual financial Statements covering the grant period, consisting of the organization’s statement of income and expense and balance sheet signed by an appropriate Official of the organization. Financial statements will be submitted within 90 days after the Grantee’s fiscal year.

#### **§§ 3570.278–3570.280 [Reserved]**

#### **§ 3570.281 Grant servicing.**

Grants will be serviced in accordance with 7 CFR part 1951, subpart E.

#### **§ 3570.282 [Reserved]**

#### **§ 3570.283 Exception authority.**

The Administrator may make an exception to any requirement or provision of this subpart, if such an exception is necessary to implement the intent of the authorizing statutes in a time of national emergency or in

accordance with a Presidentially-declared disaster, or on a case-by-case basis, when such an exception is in the best financial interest of the Federal Government and is otherwise not in conflict with applicable laws. No exceptions, however, will be granted for Applicant, Ultimate Recipient, or Project eligibility.

#### **§ 3570.284 Review or appeal rights.**

A person may seek a review of an Agency decision under this subpart from the appropriate Agency official that oversees the program in question or appeal to the USDA National Appeals Division in accordance with 7 CFR part 11.

#### **§§ 3570.285–3570.299 [Reserved]**

#### **§ 3570.300 OMB control number.**

The reporting and recordkeeping requirements contained in this regulation have been submitted to the Office of Management and Budget (OMB) for approval.

Dated: December 11, 2015.

**Tony Hernandez,**

*Administrator, Rural Housing Service.*

[FR Doc. 2016–00479 Filed 1–13–16; 8:45 am]

BILLING CODE 3410–XV–P

## **DEPARTMENT OF TRANSPORTATION**

### **Federal Aviation Administration**

#### **14 CFR Part 39**

[Docket No. FAA–2015–1275; Directorate Identifier 2014–NM–070–AD; Amendment 39–18354; AD 2015–26–06]

RIN 2120–AA64

#### **Airworthiness Directives; Airbus Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are superseding Airworthiness Directive (AD) 2004–14–09 for certain Airbus Model A320–211, –212, and –231 airplanes. AD 2004–14–09 required repetitive inspections for fatigue cracking of the lower surface panel on the wing center box, and repair if necessary; and modification of the lower surface panel on the wing center box, which constitutes terminating action for the repetitive inspections. This new AD retains the requirements of AD 2004–14–09, reduces the compliance times for the repetitive inspections, and requires an additional repair for certain airplanes. This AD was

prompted by a determination that, based on the average flight duration, the average weight of fuel at landing is higher than that defined for the analysis of the fatigue-related tasks; and that shot peening might have been improperly done on the chromic acid anodizing (CAA) protection, which would adversely affect fatigue crack protection. We are issuing this AD to detect and correct fatigue cracking of the lower surface panel on the wing center box, which could result in reduced structural integrity of the airplane.

**DATES:** This AD becomes effective February 18, 2016.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of February 18, 2016.

The Director of the Federal Register approved the incorporation by reference of certain other publications listed in this AD as of August 13, 2004 (69 FR 41398, July 9, 2004).

The Director of the Federal Register approved the incorporation by reference of certain other publications listed in this AD as of November 27, 1998 (63 FR 56542, October 22, 1998).

**ADDRESSES:** You may examine the AD docket on the Internet at <http://www.regulations.gov/#!docketDetail;D=FAA-2015-1275>; or in person at the Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC.

For service information identified in this final rule, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-1275.

**FOR FURTHER INFORMATION CONTACT:** Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1405; fax 425-227-1149.

**SUPPLEMENTARY INFORMATION:**

## Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to supersede AD 2004-14-09, Amendment 39-13718 (69 FR 41398, July 9, 2004). AD 2004-14-09 applied to certain Airbus Model A320-211, -212, and -231 airplanes. The NPRM published in the **Federal Register** on May 8, 2015 (80 FR 26492). The NPRM was prompted by a determination that, based on the average flight duration, the average weight of fuel at landing is higher than that defined for the analysis of the fatigue-related tasks; and that shot peening might have been improperly done on the CAA protection, which would adversely affect fatigue crack protection. The NPRM proposed to continue to require repetitive inspections for fatigue cracking of the lower surface panel on the wing center box, and repair if necessary; and modification of the lower surface panel on the wing center box, which constitutes terminating action for the repetitive inspections. The NPRM also proposed to reduce the compliance times for the repetitive inspections, and would require a repair for certain airplanes. We are issuing this AD to detect and correct fatigue cracking of the lower surface panel on the wing center box, which could result in reduced structural integrity of the airplane.

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2014-0065, dated March 14, 2014 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition on certain Model A320-211, -212, and -231 airplanes. The MCAI states:

During center fuselage certification full scale test, damage was found in the center wing box (CWB) lower surface panel.

This condition, if not detected and corrected, could affect the structural integrity of the CWB.

To prevent such damage, Airbus developed mod 22418 which consists in shot-peening of the lower panel in the related area. Mod 22418 has been embodied in production from aeroplane [manufacturer serial number] (MSN) 0359. For unmodified in-service aeroplanes, Airbus issued Service Bulletin (SB) A320-57-1082 to introduce repetitive High Frequency Eddy Current (HFEC) inspections on the external face of the center wing box lower panel between Frame (FR) 41 and FR42 to detect damage.

DGAC [Direction Générale de l'Aviation Civile] France issued AD 2002-342 [[http://ad.easa.europa.eu/blob/20023420tb\\_superseded.pdf](http://ad.easa.europa.eu/blob/20023420tb_superseded.pdf)]/AD\_F-2002-342\_1 to require these inspections and, depending on findings, applicable corrective action(s).

Airbus also issued SB A320-57-1043 as an optional terminating action for the repetitive inspections required by DGAC France AD 2002-342.

Since that [DGAC] AD was issued, the results of a survey, carried out on the A320 fleet, highlighted some differences between the mission parameters, mainly on the weight of fuel at landing and on the average flight duration, which are higher than those defined for the analysis of the fatigue related tasks.

These findings have led to an adjustment of the A320 reference fatigue mission. Consequently, the threshold and intervals of these repetitive inspections have been revised and a new threshold figure expressed in flight hours (FH) has been established.

In addition, it has been identified that, on aeroplanes that have been modified in accordance with Airbus SB A320-57-1043 (Airbus mod 22418) at Revision 05 or an earlier Revision, the shot peening may have been improperly done on the Chromic Acid Anodizing (CAA) protection, which has no fatigue benefit effect. Therefore, the inspections per Airbus SB A320-57-1082 are required again on these aeroplanes.

Consequently, new shot-peening procedures with proper CAA protection removal instructions have been developed and their embodiment through Airbus SB A320-57-1043 Revision 06 cancels the repetitive inspections per Airbus SB A320-57-1082, as required by DGAC France AD 2002-342.

For the reasons described above, this new [EASA] AD retains the requirements of DGAC France AD 2002-342, which is superseded, but requires these actions to be accomplished within reduced thresholds and intervals. In addition, the optional terminating action provision (SB A320-57-1043) is amended by including reference to the SB at Revision 06.

The optional terminating action described in Airbus Service Bulletin A320-57-1043, Revision 06, dated December 5, 2013, is accomplishing shot peening in the radius of the milling step between stiffeners 13 and 14 near the fuel pump aperture. You may examine the MCAI in the AD docket on the Internet at <http://www.regulations.gov/#!documentDetail;D=FAA-2015-1275-0002>.

## Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (80 FR 26492, May 8, 2015) or on the determination of the cost to the public.

## Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting this AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the NPRM (80 FR

26492, May 8, 2015) for correcting the unsafe condition; and

- Do not add any additional burden upon the public than was already proposed in the NPRM (80 FR 26492, May 8, 2015).

#### **Related Service Information Under 1 CFR Part 51**

Airbus has issued Service Bulletin A320–57–1043, Revision 06, dated December 5, 2013. This service information describes procedures for shot peening in the radius of the milling step between stiffeners 13 and 14 near the fuel pump aperture.

Airbus has also issued Service Bulletin A320–57–1082, Revision 04, dated December 5, 2013. This service information describes procedures for inspections for cracking of the lower surface panel on the wing center box.

This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the ADDRESSES section.

#### **Costs of Compliance**

We estimate that this AD affects 46 airplanes of U.S. registry.

The actions that were required by AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), and retained in this AD take about 25 work-hours per product, at an average labor rate of \$85 per work-hour. Based on these figures, the estimated cost of the actions that were required by AD 2004–14–09 is \$2,125 per product.

The new requirements of this AD will add no additional economic burden.

We have received no definitive data that will enable us to provide cost estimates for the on-condition actions specified in this AD. We have no way of determining the number of aircraft that might need these actions.

#### **Authority for This Rulemaking**

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority

because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

#### **Regulatory Findings**

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

1. Is not a "significant regulatory action" under Executive Order 12866;
2. Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
3. Will not affect intrastate aviation in Alaska; and
4. Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

#### **Examining the AD Docket**

You may examine the AD docket on the Internet at <http://www.regulations.gov/#/docketDetail;D=FAA-2015-1275>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The street address for the Docket Operations office (telephone 800–647–5527) is in the ADDRESSES section.

#### **List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### **Adoption of the Amendment**

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

#### **PART 39—AIRWORTHINESS DIRECTIVES**

- 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

##### **§ 39.13 [Amended]**

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), and adding the following new AD:

**2015–26–06 Airbus:** Amendment 39–18354. Docket No. FAA–2015–1275; Directorate Identifier 2014–NM–070–AD.

##### **(a) Effective Date**

This AD becomes effective February 18, 2016.

##### **(b) Affected ADs**

This AD replaces AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004).

##### **(c) Applicability**

This AD applies to Airbus Model A320–211, –212, and –231 airplanes, certificated in any category, all manufacturer serial numbers, except those on which Airbus Modification 22418 has been embodied in production.

##### **(d) Subject**

Air Transport Association (ATA) of America Code 57, Wings.

##### **(e) Reason**

This AD was prompted by a determination that, based on the average flight duration, the average weight of fuel at landing is higher than that defined for the analysis of the fatigue-related tasks; and that shot peening might have been improperly done on the chromic acid anodizing (CAA) protection, which would adversely affect fatigue crack protection. We are issuing this AD to detect and correct fatigue cracking of the lower surface panel on the wing center box (WCB), which could result in reduced structural integrity of the airplane.

##### **(f) Compliance**

Comply with this AD within the compliance times specified, unless already done.

##### **(g) Retained Repetitive Inspections, With No Changes**

This paragraph restates the requirements of paragraph (a) of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), with no changes. Except as provided by paragraph (k) of this AD: Prior to the accumulation of 20,000 total flight cycles, or within 60 days after November 27, 1998 (the effective date of AD 98–22–05, Amendment 39–10851 (63 FR 56542, October 22, 1998)), whichever occurs later, perform a high frequency eddy current (HFEC) inspection to detect fatigue cracking of the lower surface panel on the WCB, in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997; or Revision 03, dated April 30, 2002. Repeat the HFEC inspection thereafter at intervals not to exceed 7,500 flight cycles until the actions required by paragraph (i) of this AD are accomplished.

##### **(h) Retained Repair, With No Changes**

This paragraph restates the requirements of paragraph (b) of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), with no changes. Except as provided by paragraph (j) of this AD, if any cracking is detected during any inspection required by paragraph (g) of this AD: Prior to further flight, repair in accordance with Airbus

Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997; or Revision 03, dated April 30, 2002. Accomplishment of the repair constitutes terminating action for the repetitive inspections required by paragraph (g) of this AD for the repaired area only.

**(i) Retained Inspection/Modification/Repair, With New Terminating Action**

This paragraph restates the requirements of paragraph (c) of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), with new terminating action provided. Prior to the accumulation of 25,000 total flight cycles, or within 60 days after November 27, 1998 (the effective date of AD 98–22–05, Amendment 39–10851 (63 FR 56542, October 22, 1998)), whichever occurs later: Perform an HFEC inspection to detect fatigue cracking of the lower surface panel on the WCB, in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997; or Revision 03, dated April 30, 2002. Accomplishment of the initial inspection required by paragraph (p) of this AD constitutes terminating action for the inspection requirements of this paragraph.

(1) If no cracking is detected: Prior to further flight, modify the lower surface panel on the WCB, in accordance with Airbus Service Bulletin A320–57–1043, Revision 02, dated May 14, 1997; or Revision 05, dated April 30, 2002. Accomplishment of the modification constitutes terminating action for the requirements of paragraph (g) of this AD.

(2) Except as provided by paragraph (j) of this AD: If any cracking is detected, prior to further flight, repair in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997, or Revision 03, dated April 30, 2002; and modify any uncracked area, in accordance with Airbus Service Bulletin A320–57–1043, Revision 02, dated May 14, 1997, or Revision 05, dated April 30, 2002. Accomplishment of the repair of cracked area(s) and modification of uncracked area(s) constitutes terminating action for the requirements of paragraph (g) of this AD.

**(j) Retained Service Bulletin Exception, With Revised Repair Instructions**

This paragraph restates the requirements of paragraph (d) of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), with revised repair instructions. If any cracking is detected during any inspection required by paragraph (h) or (i)(2) of this AD, and the applicable service bulletin specifies to contact Airbus for an appropriate action: Prior to further flight, repair using a method approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate; or the Direction Générale de l'Aviation Civile (DGAC) (or its delegated agent); or the European Aviation Safety Agency (EASA); or Airbus's EASA Design Organization Approval (DOA). After the effective date of this AD only repair using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or the EASA; or Airbus's EASA DOA.

**(k) Retained Provision for Certain Inspection Exception, With No Changes**

This paragraph restates the provision of paragraph (e) of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), with no changes. The actions required by paragraph (g) of this AD are not required to be accomplished if the requirements of paragraph (i) of this AD are accomplished at the time specified in paragraph (g) of this AD.

**(l) Retained Initial Inspection, With Terminating Action**

This paragraph restates the requirements of paragraph (f) of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), with terminating action provided. For airplanes on which neither the inspection required by paragraph (g) of this AD nor the modification required by paragraph (i)(1) of this AD has been done before August 13, 2004 (the effective date of AD 2004–14–09): Perform an HFEC inspection to detect fatigue cracking of the lower surface panel on the WCB, in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997; or Revision 03, dated April 30, 2002; at the later of the times specified in paragraphs (l)(1) and (l)(2) of this AD. Accomplishment of the inspection required by this paragraph terminates the requirements of paragraph (g) of this AD. Accomplishment of the initial inspection required by paragraph (p) of this AD terminates the inspection requirements of this paragraph.

(1) Prior to the accumulation of 13,200 total flight cycles or 39,700 total flight hours, whichever is first.

(2) Prior to the accumulation of 20,000 total flight cycles, or within 3,500 flight cycles after August 13, 2004 (the effective date of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004)), whichever is later.

**(m) Retained Repetitive Inspections, With No Changes**

This paragraph restates the requirements of paragraph (g) of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), with no changes. If no cracking is detected during the inspection required by paragraph (g) or (l) of this AD: Repeat the inspection required by paragraph (l) of this AD at the applicable time specified in paragraph (m)(1) or (m)(2) of this AD. Accomplishment of the modification required by paragraph (i)(1) of this AD terminates the requirements of this paragraph.

(1) For airplanes on which the inspections required by paragraph (g) of this AD have been initiated before August 13, 2004 (the effective date of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004)): Do the next inspection within 5,700 flight cycles after accomplishment of the last inspection, or within 1,800 flight cycles after August 13, 2004, whichever is later. Repeat the inspection thereafter at intervals not to exceed 5,700 flight cycles.

(2) For airplanes on which no inspection required by paragraph (g) of this AD has been done before August 13, 2004 (the effective date of AD 2004–14–09, Amendment 39–

13718 (69 FR 41398, July 9, 2004)): Do the next inspection within 5,700 flight cycles after accomplishment of the inspection required by paragraph (l) of this AD. Repeat the inspection thereafter at intervals not to exceed 5,700 flight cycles.

**(n) Retained Repair/Modification, With Revised Repair Instructions**

This paragraph restates the requirements of paragraph (h) of AD 2004–14–09, Amendment 39–13718 (69 FR 41398, July 9, 2004), with revised repair instructions. If any cracking is detected during any inspection required by paragraph (l) or (m) of this AD, prior to further flight, repair in accordance with Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997, or Revision 03, dated April 30, 2002; and modify any uncracked area, in accordance with Airbus Service Bulletin A320–57–1043, Revision 02, dated May 14, 1997, or Revision 05, dated April 30, 2002. Where Airbus Service Bulletin A320–57–1082 specifies to contact Airbus for an appropriate repair action: Prior to further flight, repair using a method approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate; or the DGAC (or its delegated agent); or EASA; or Airbus's EASA DOA. After the effective date of this AD only repair using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; or EASA; or Airbus's EASA DOA. Accomplishment of the repair of cracked area(s) and modification of uncracked area(s) constitutes terminating action for the requirements of paragraphs (g) through (n) of this AD.

**(o) New Requirement of This AD: Repair of Certain Airplanes**

For airplanes on which the actions specified in Airbus Service Bulletin A320–57–1043 have not been accomplished, and on which a repair has been accomplished, as specified in the service information identified in paragraph (o)(1), (o)(2), (o)(3), or (o)(4) of this AD: Within 30 days after the effective date of this AD, repair using a method approved by the Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA; the EASA; or Airbus's EASA DOA.

(1) Airbus Service Bulletin A320–57–1082, dated October 31, 1996.

(2) Airbus Service Bulletin A320–57–1082, Revision 01, dated December 10, 1997.

(3) Airbus Service Bulletin A320–57–1082, Revision 02, dated July 26, 1999.

(4) Airbus Service Bulletin A320–57–1082, Revision 03, dated April 30, 2002.

**(p) New Requirement of This AD: Repetitive WCB Inspections**

At the applicable time specified in paragraphs (p)(1) and (p)(2) of this AD: Do an HFEC inspection for cracking of the lower surface panel on the WCB, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320–57–1082, Revision 04, dated December 5, 2013. Repeat the inspection of the lower surface panel on the WCB thereafter at intervals not to exceed 7,200 flight cycles or 14,400 flight hours, whichever occurs first. Accomplishment of

the initial inspection required by this paragraph terminates the inspections required by paragraphs (g), (i), and (l) of this AD.

(1) For airplanes on which the actions specified in Airbus Service Bulletin A320-57-1043 have not been done: At the later of the times specified in paragraphs (p)(1)(i) and (p)(1)(ii) of this AD.

(i) Before the accumulation of 20,700 flight cycles or 41,400 flight hours, whichever occurs first since first flight of the airplane.

(ii) Within 7,200 flight cycles or 14,400 flight hours, whichever occurs first after doing the most recent inspection as specified in the service information specified in paragraph (o)(1), (o)(2), (o)(3), or (o)(4) of this AD.

(2) For airplanes on which the actions specified in Airbus Service Bulletin A320-57-1043 have been done: At the latest of the times specified in paragraphs (p)(2)(i), (p)(2)(ii), and (p)(2)(iii) of this AD.

(i) Within 7,200 flight cycles or 14,400 flight hours, whichever occurs first since doing the actions specified in Airbus Service Bulletin A320-57-1043.

(ii) Within 3,750 flight cycles or 7,500 flight hours, whichever occurs first after July 31, 2012 (as described in Airbus Service Bulletin A320-57-1082, Revision 04, dated December 5, 2013).

(iii) Within 850 flight cycles or 1,700 flight hours, whichever occurs first after the effective date of this AD.

#### (q) New Requirement of This AD: Repair of WCB

If any crack is found during any inspection required by paragraph (p) of this AD: Before further flight, repair using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; the EASA; or Airbus's EASA DOA.

#### (r) New Optional Terminating Action

Modification of an airplane, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A320-57-1043, Revision 06, dated December 5, 2013, constitutes terminating action for the actions required by paragraph (p) of this AD.

#### (s) Credit for Previous Actions

This paragraph provides credit for applicable actions required by paragraphs (g) through (n) of this AD, if those actions were performed before the effective date of this AD using the applicable Airbus Service Information provided in paragraphs (s)(1) through (s)(8) of this AD.

(1) Airbus Service Bulletin A320-57-1043, dated February 16, 1993, which is not incorporated by reference in this AD.

(2) Airbus Service Bulletin A320-57-1043, Revision 01, dated June 14, 1996, which is not incorporated by reference in this AD.

(3) Airbus Service Bulletin A320-57-1043, Revision 02, dated May 14, 1997, which was incorporated by reference on November 27, 1998 (63 FR 56542, October 22, 1998).

(4) Airbus Service Bulletin A320-57-1043, Revision 03, dated October 24, 1997, which is not incorporated by reference in this AD.

(5) Airbus Service Bulletin A320-57-1043, Revision 04, dated May 15, 1999, which is not incorporated by reference in this AD.

(6) Airbus Service Bulletin A320-57-1082, Revision 01, dated December 10, 1997, which was incorporated by reference on November 27, 1998 (63 FR 56542, October 22, 1998).

(7) Airbus Service Bulletin A320-57-1082, Revision 02, dated July 26, 1999, which is not incorporated by reference in this AD.

(8) Airbus Service Bulletin A320-57-1082, Revision 03, dated April 30, 2002, which was incorporated by reference on August 13, 2004 (69 FR 41398, July 9, 2004).

#### (t) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs)*: The Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the International Branch, send it to ATTN: Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1405; fax 425-227-1149. Information may be emailed to: [9-ANM-116-AMOC-REQUESTS@faa.gov](mailto:9-ANM-116-AMOC-REQUESTS@faa.gov). Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(2) *Contacting the Manufacturer*: As of the effective date of this AD, for any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, International Branch, ANM-116, Transport Airplane Directorate, FAA; or the EASA; or Airbus's EASA DOA. If approved by the DOA, the approval must include the DOA-authorized signature.

#### (u) Related Information

(1) Refer to Mandatory Continuing Airworthiness Information (MCAI) EASA Airworthiness Directive 2014-0065, dated March 14, 2014, for related information. This MCAI may be found in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-1275.

(2) Service information identified in this AD that is not incorporated by reference is available at the addresses specified in paragraphs (v)(6) and (v)(7) of this AD.

#### (v) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless this AD specifies otherwise.

(3) The following service information was approved for IBR on February 18, 2016.

(i) Airbus Service Bulletin A320-57-1043, Revision 06, dated December 5, 2013.

(ii) Airbus Service Bulletin A320-57-1082, Revision 04, dated December 5, 2013. (4) The following service information was approved for IBR on August 13, 2004 (69 FR 41398, July 9, 2004).

(i) Airbus Service Bulletin A320-57-1043, Revision 05, dated April 30, 2002.

(ii) Airbus Service Bulletin A320-57-1082, Revision 03, dated April 30, 2002.

(5) The following service information was approved for IBR on November 27, 1998 (63 FR 56542, October 22, 1998).

(i) Airbus Service Bulletin A320-57-1043, Revision 02, dated May 14, 1997. Pages 1 through 6, 8, 13, and 14 of this service bulletin are marked Revision 02, dated May 14, 1997; pages 7, 9 through 12, and 15 of this service bulletin are from the original issue, dated February 16, 1993.

(ii) Airbus Service Bulletin A320-57-1082, Revision 01, dated December 10, 1997.

(6) For service information identified in this AD, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet <http://www.airbus.com>.

(7) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(8) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on December 18, 2015.

**Jeffrey E. Duven,**

*Manager, Transport Airplane Directorate, Aircraft Certification Service.*

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## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

**[Docket No. FAA-2015-1982; Directorate Identifier 2014-NM-108-AD; Amendment 39-18353; AD 2015-26-05]**

**RIN 2120-AA64**

#### **Airworthiness Directives; Fokker Services B.V. Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) for all Fokker Services B.V. Model F.28 Mark