

for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2016-09-04 Bombardier, Inc.: Amendment 39-18502; Docket No. FAA-2015-4814; Directorate Identifier 2015-NM-105-AD.

(a) Effective Date

This AD is effective June 6, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to Bombardier, Inc. Model CL-600-2B19 (Regional Jet Series 100 & 440) airplanes, certificated in any category, serial

numbers 7003 through 7067 inclusive, 7069 through 7990 inclusive, and 8000 through 8999 inclusive.

(d) Subject

Air Transport Association (ATA) of America Code 27, Flight Controls.

(e) Reason

This AD was prompted by the discovery of a number of incorrectly calibrated angle of attack (AOA) transducers installed in the stall protection system. We are issuing this AD to detect and replace incorrectly calibrated AOA transducers; incorrect calibration of the transducers could result in late activation of the stick pusher.

(f) Compliance

Comply with this AD within the compliance times specified, unless already done.

(g) Replacement

For AOA transducers identified in paragraph 1.A., "Effectivity," of Bombardier Service Bulletin 601R-27-164, dated March 30, 2015: Within 2,500 flight hours or 12 months, whichever occurs first after the effective date of this AD, replace the AOA transducers with correctly calibrated AOA transducers, in accordance with the Accomplishment Instructions of Bombardier Service Bulletin 601R-27-164, dated March 30, 2015.

(h) Parts Installation Prohibition

As of the effective date of this AD, no person may install, on any airplane, an AOA transducer having a part number or serial number listed in paragraph 1.A., "Effectivity," of Bombardier Service Bulletin 601R-27-164, dated March 30, 2015.

(i) Other FAA AD Provisions

The following provisions also apply to this AD:

(1) Alternative Methods of Compliance (AMOCs): The Manager, New York Aircraft Certification Office (ACO), ANE-170, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the ACO, send it to ATTN: Program Manager, Continuing Operational Safety, FAA, New York ACO, 1600 Stewart Avenue, Suite 410, Westbury, NY 11590; telephone 516-228-7300; fax 516-794-5531. Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office. The AMOC approval letter must specifically reference this AD.

(2) Contacting the Manufacturer: For any requirement in this AD to obtain corrective actions from a manufacturer, the action must be accomplished using a method approved by the Manager, New York ACO, ANE-170, FAA; or Transport Canada Civil Aviation (TCCA); or Bombardier, Inc.'s TCCA Design Approval Organization (DAO). If approved by the DAO, the approval must include the DAO-authorized signature.

(j) Related Information

For more information about this AD, contact Cesar Gomez, Aerospace Engineer, Airframe and Mechanical Systems Branch, ANE-171, FAA, New York ACO, 1600 Stewart Avenue, Suite 410, Westbury, NY 11590; telephone 516-228-7318; fax 516-794-5531.

(k) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Bombardier Service Bulletin 601R-27-164, dated March 30, 2015.

(ii) Reserved.

(3) For service information identified in this AD, contact Bombardier, Inc., 400 Côte-Vertu Road West, Dorval, Québec H4S 1Y9, Canada; telephone 514-855-5000; fax 514-855-7401; email thd.crj@aero.bombardier.com; Internet <http://www.bombardier.com>.

(4) You may view this service information at FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on April 20, 2016.

John P. Piccola, Jr.,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-09791 Filed 4-29-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-3970; Directorate Identifier 2015-SW-006-AD; Amendment 39-18497; AD 2016-08-20]

RIN 2120-AA64

Airworthiness Directives; Airbus Helicopters (Previously Eurocopter France)

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are superseding airworthiness directive (AD) 2014-12-51 for Airbus Helicopters (previously Eurocopter France) Model EC130B4 and

EC130T2 helicopters. AD 2014–12–51 required repetitively inspecting the tailboom to Fenestron junction frame (junction frame) for a crack. This new AD retains the requirements of AD 2014–12–51, changes the applicability from helicopters with certain hours time-in-service (TIS) to junction frames with certain hours TIS, and adds a compliance time for sling cycles to the junction frame inspection interval. The actions of this AD are intended to detect a crack and to prevent failure of the junction frame, which could result in loss of the Fenestron and subsequent loss of control of the helicopter.

DATES: This AD is effective June 6, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain document listed in this AD as of June 6, 2016.

ADDRESSES: For service information identified in this final rule, contact Airbus Helicopters, Inc., 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641–0000 or (800) 232–0323; fax (972) 641–3775; or at <http://www.airbushelicopters.com/techpub>. You may review the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy, Room 6N–321, Fort Worth, TX 76177. It is also on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2015–3970.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA–2015–3970; or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the European Aviation Safety Agency (EASA) AD, any incorporated-by-reference service information, the economic evaluation, any comments received, and other information. The street address for the Docket Operations Office (phone: 800–647–5527) is U.S. Department of Transportation, Docket Operations Office, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Robert Grant, Aviation Safety Engineer, Safety Management Group, FAA, Rotorcraft Directorate, FAA, 10101 Hillwood Pkwy, Fort Worth, TX 76177; email robert.grant@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

On September 25, 2015, at 80 FR 57742, the **Federal Register** published our notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to remove AD 2014–12–51, Amendment 39–17921 (79 FR 45335, August 5, 2014), and add a new AD. AD 2014–12–51 applied to Airbus Helicopters Model EC130B4 and EC130T2 helicopters with 690 or more hours TIS and required, within 10 hours TIS, dye-penetrant inspecting certain areas of the junction frame for a crack. AD 2014–12–51 also required, at intervals not exceeding 25 hours TIS, either repeating the dye-penetrant inspection or performing a borescope inspection of certain areas of the junction frame for a crack. If there was a crack, AD 2014–12–51 required replacing the junction frame. AD 2014–12–51 was prompted by two incidents of crack propagation through the junction frame that initiated in the lower right-hand side between the web and the flange where the lower spar of the tailboom is joined. The cracks were significant in length and not visible from the outside of the helicopter.

The NPRM was prompted by AD No. 2015–0033–E dated February 24, 2015 (AD 2015–0033–E), issued by EASA, which is the Technical Agent for the Member States of the European Union, to correct an unsafe condition on Airbus Helicopters EC130B4 and EC130T2 helicopters. The NPRM proposed to require retaining the dye penetrant and borescope inspections in AD 2014–12–51 but with revised compliance times. The NRPM also proposed to change the applicability to helicopters with 690 hours TIS accumulated on the junction frame instead of on the helicopter, and proposed including an inspection interval defined in sling cycles. These actions were intended to detect a crack and to prevent failure of the junction frame, which could result in loss of the Fenestron and subsequent loss of control of the helicopter.

Comments

After our NPRM (80 FR 57742, September 25, 2015) was published, we received a comment from one commenter.

Request

One commenter requested the addition of a 10-hour or 250-sling cycle visual pilot check for helicopters with Modification 350A087421 or that have complied with Airbus Helicopters Service Bulletin No. EC130–53–029, Revision 0, dated February 20, 2015 (SB EC130–53–029). The commenter stated

this pilot check would benefit operators and provide the same level of safety.

We disagree. While the EASA AD allows the check requested by the commenter as an alternative method, because the cause of the fatigue cracking is still under investigation, we cannot determine that this method would correct the unsafe condition.

FAA's Determination

These helicopters have been approved by the aviation authority of France and are approved for operation in the United States. Pursuant to our bilateral agreement with France, EASA, its technical representative, has notified us of the unsafe condition described in the EASA AD. We are issuing this AD because we evaluated all information provided by EASA, reviewed the relevant information, considered the comment received, and determined the unsafe condition exists and is likely to exist or develop on other helicopters of the same type designs and that air safety and the public interest require adopting the AD requirements as proposed.

Interim Action

We consider this AD to be an interim action. If final action is later identified, we might consider further rulemaking then.

Differences Between This AD and the EASA AD

The EASA AD includes alternate compliance instructions for helicopters modified with a cut-out in production by Airbus Helicopters Modification 350A087421 or in service by compliance with SB EC130–53–029. This AD does not.

Related Service Information Under 1 CFR Part 51

We reviewed Airbus Helicopters Emergency Alert Service Bulletin No. 05A017, Revision 2, dated February 20, 2015 (EASB 05A017), for Model EC130B4 and EC130T2 helicopters. EASB 05A017 describes alternate procedures for inspecting outside the tailboom for a crack at reduced inspection intervals in combination with the internal inspections at extended intervals. EASB 05A017 also specifies adding sling cycles to the existing flight hour inspection interval for helicopters that perform external load-carrying operations. EASA issued AD No. 2015–0033–E mandating the requirements in EASB 05A017 to ensure the continued airworthiness of these helicopters.

This service information is reasonably available because the interested parties have access to it through their normal

course of business or by the means identified in the **ADDRESSES** section.

Other Related Service Information

Airbus Helicopters also issued SB EC130–53–029, which contains procedures to cut out the skin and splice at the junction frame to facilitate the external inspection specified in EASB 05A017.

Costs of Compliance

We estimate that this AD affects 208 helicopters of U.S. Registry. We estimate that operators may incur the following costs in order to comply with this AD. At an average labor rate of \$85 per work-hour, dye-penetrant inspecting the junction frame will require 1 work-hour, for a cost of \$85 per helicopter and a total cost of \$17,680 for the U.S. fleet, per inspection cycle. Borescope inspecting the junction frame will require 0.5 work-hour, for a cost of \$43 per helicopter and a total cost of \$8,944 for the U.S. fleet, per inspection cycle.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on helicopters identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);

(3) Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction; and

(4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by removing Airworthiness Directive (AD) 2014–12–51, Amendment 39–17921 (79 FR 45335, August 5, 2014), and adding the following new AD:

2016–08–20 Airbus Helicopters (Previously Eurocopter France): Amendment 39–18497; Docket No. FAA–2015–3970; Directorate Identifier 2015–SW–006–AD.

(a) Applicability

This AD applies to Airbus Helicopters Model EC130B4 and EC130T2 helicopters with a tailboom to fenestron junction frame (junction frame) that has 690 or more hours time-in-service (TIS), certificated in any category.

(b) Unsafe Condition

This AD defines the unsafe condition as a crack in the junction frame. This condition could result in failure of the junction frame, which could result in loss of the Fenestron and subsequent loss of control of the helicopter.

(c) Affected ADs

This AD supersedes AD 2014–12–51, Amendment 39–17921 (79 FR 45335, August 5, 2014).

(d) Effective Date

This AD becomes effective June 6, 2016.

(e) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

(f) Required Actions

(1) Before the junction frame reaches 700 hours TIS or within 10 hours TIS, whichever occurs later, remove the horizontal stabilizer, clean the junction frame, and dye-penetrant inspect around the circumference of the junction frame for a crack in the areas shown in Figure 1 of Airbus Helicopters EC130 Emergency Alert Service Bulletin No. 05A017, Revision 2, dated February 20, 2015 (EASB 05A017). Pay particular attention to the area around the 4 spars (item b) of Figure 1 of EASB 05A017. An example of a crack is shown in Figure 3 of EASB 05A017.

(2) Within 25 hours TIS or 390 sling cycles, whichever occurs first after the inspection required by paragraph (f)(1) of this AD, and thereafter at intervals not exceeding 25 hours TIS or 390 sling cycles, whichever occurs first, either perform the actions of paragraph (f)(1) of this AD or, if the area is clean, using a borescope, inspect around the circumference of the junction frame for a crack in the areas shown in Figure 2 of EASB 05A017. Pay particular attention to the area around the 4 spars (item b) of Figure 2 of EASB 05A017. An example of a crack is shown in Figure 3 of EASB 05A017. For purposes of this AD, a sling cycle is defined as one landing with or without stopping the rotor or one external load-carrying operation; an external load-carrying operation occurs each time a helicopter picks up an external load and drops it off.

(3) If there is a crack, before further flight, replace the junction frame.

(g) Special Flight Permits

Special flight permits are prohibited.

(h) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Safety Management Group, FAA, may approve AMOCs for this AD. Send your proposal to: Robert Grant, Aviation Safety Engineer, Safety Management Group, Rotorcraft Directorate, FAA, 10101 Hillwood Pkwy., Fort Worth, TX 76177; telephone (817) 222–5110; email 9-ASW-FTW-AMOC-Requests@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office, before operating any aircraft complying with this AD through an AMOC.

(i) Additional Information

(1) Airbus Helicopters Service Bulletin No. EC130–53–029, Revision 0, dated February 20, 2015, which is not incorporated by reference, contains additional information about the subject of this final rule. For service information identified in this final rule, contact Airbus Helicopters, Inc., 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641–0000 or (800) 232–0323; fax (972) 641–3775; or at <http://www.airbushelicopters.com/techpub>. You may review a copy of the service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy., Room 6N–321, Fort Worth, TX 76177.

(2) The subject of this AD is addressed in European Aviation Safety Agency (EASA) AD No. 2015-0033-E, dated February 24, 2015. You may view the EASA AD on the Internet at <http://www.regulations.gov> in Docket No. FAA-2015-3970.

(j) Subject

Joint Aircraft Service Component (JASC)
Code: 5302: Rotorcraft Tailboom.

(k) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Airbus Helicopters Emergency Alert Service Bulletin No. 05A017, Revision 2, dated February 20, 2015.

(ii) Reserved.

(3) For Airbus Helicopters service information identified in this final rule, contact Airbus Helicopters, Inc., 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>.

(4) You may view this service information at FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy., Room 6N-321, Fort Worth, TX 76177. For information on the availability of this material at the FAA, call (817) 222-5110.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call (202) 741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Fort Worth, Texas, on April 15, 2016.

Scott A. Horn,

*Acting Manager, Rotorcraft Directorate,
Aircraft Certification Service.*

[FR Doc. 2016-09235 Filed 4-29-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2014-0338; Directorate Identifier 2014-CE-010-AD; Amendment 39-18495; AD 2016-08-18]

RIN 2120-AA64

Airworthiness Directives; Piper Aircraft, Inc. Airplanes

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain

Piper Aircraft, Inc. Model PA-31-350 airplanes. This AD was prompted by a report of an engine fire caused by a leak in the fuel pump inlet hose. This AD requires inspecting the fuel hose assembly and the turbocharger support assembly for proper clearance between them, inspecting each assembly for any sign of damage, and making any necessary repairs or replacements. We are issuing this AD to correct the unsafe condition on these products.

DATES: This AD is effective June 6, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of June 6, 2016.

ADDRESSES: For service information identified in this final rule, contact Piper Aircraft, Inc., 2926 Piper Drive, Vero Beach, Florida 32960; telephone: (772) 567-4361; fax: (772) 978-6573; Internet: www.piper.com/home/pages/Publications.cfm. You may view this referenced service information at the FAA, Small Airplane Directorate, 901 Locust, Kansas City, Missouri 64106. For information on the availability of this material at the FAA, call (816) 329-4148. It is also available on the internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2014-0338.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2014-0338; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Gary Wechsler, Aerospace Engineer, FAA, Atlanta Aircraft Certification Office, 1701 Columbia Avenue, College Park, Georgia 30337; telephone: (404) 474-5575; fax: (404) 474-5606; email: gary.wechsler@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a supplemental notice of proposed rulemaking (SNPRM) to amend 14 CFR part 39 by adding an AD that would apply to certain Piper Aircraft, Inc. Model PA-31-350

airplanes. The SNPRM published in the **Federal Register** on January 26, 2016 (81 FR 4214). We preceded the SNPRM with a notice of proposed rulemaking (NPRM) that published in the **Federal Register** on June 3, 2014 (79 FR 31888). The NPRM proposed to require inspecting the fuel hose assembly and the turbocharger support assembly for proper clearance between them, inspecting each assembly for any sign of damage, and making any necessary repairs or replacements. The NPRM was prompted by a report of an engine fire on a Piper Aircraft, Inc. (Piper) Model PA-31-350 airplane. Investigation revealed that the fire was caused by a leak in the fuel pump inlet hose that resulted from repeated contact with an adjacent turbocharger support assembly caused by inadequate clearance between the two assemblies. The SNPRM proposed to require the same actions as proposed in the NPRM using revised service information issued by the manufacturer to clarify which engines are part of the airplane applicability and to revise the instructions for accomplishing the proposed actions.

This condition, if not corrected, could result in damage to the fuel inlet hose assembly, which could cause the fuel pump inlet hose to fail and leak fuel in the engine compartment. This condition could also cause damage to the turbocharger support assembly, which could require the turbocharger support assembly to be repaired or replaced.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the SNPRM (81 FR 4214, January 26, 2016) or on the determination of the cost to the public.

Conclusion

We reviewed the relevant data and determined that air safety and the public interest require adopting this AD as proposed except for minor editorial changes. We have determined that these minor changes:

- Are consistent with the intent that was proposed in the SNPRM (81 FR 4214, January 26, 2016) for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the SNPRM (81 FR 4214, January 26, 2016).

Related Service Information Under 1 CFR Part 51

We reviewed Piper Aircraft, Inc. Service Bulletin No. 1257A, dated August 4, 2015. The service information describes procedures for the following. This service information is reasonably