

717-55A0012, dated June 12, 2015, except as required by paragraph (i)(2) of this AD. Do all applicable related investigative and corrective actions before further flight.

Repeat the applicable inspection thereafter at the intervals specified in paragraph 1.E., "Compliance," of Boeing Alert Service Bulletin 717-55A0012, dated June 12, 2015.

(1) Do detailed inspections for any loose and missing fasteners of the vertical stabilizer leading edge as specified in "Part 4" of Boeing Alert Service Bulletin 717-55A0012, dated June 12, 2015.

(2) Do eddy current testing high frequency (ETHF) and radiographic testing (RT) inspections for any crack of the vertical stabilizer spar cap as specified in "Part 2" of Boeing Alert Service Bulletin 717-55A0012, dated June 12, 2015; or do ETHF inspections for any crack of the vertical stabilizer spar cap as specified in "Part 3" of Boeing Alert Service Bulletin 717-55A0012, dated June 12, 2015.

(i) Exceptions to the Service Information

(1) Where Boeing Alert Service Bulletin 717-55A0012, dated June 12, 2015 specifies a compliance time "after the original issue date of this service bulletin," this AD requires compliance within the specified compliance time after the effective date of this AD.

(2) If any crack is found during any inspection required by this AD, and Boeing Alert Service Bulletin 717-55A0012, dated June 12, 2015, specifies to contact Boeing for appropriate action: Before further flight, repair using a method approved in accordance with the procedures specified in paragraph (k) of this AD.

(j) Credit for Previous Actions

This paragraph provides credit for the initial inspection specified in paragraph (g) of this AD, if that inspection was performed before the effective date of this AD using Boeing MOM-MOM-14-0437-01B(R1), dated July 3, 2014, which is not incorporated by reference in this AD.

(k) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Los Angeles Aircraft Certification Office (ACO), FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the attention of the person identified in paragraph (l) of this AD. Information may be emailed to: 9-ANM-LAACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair, modification, or alteration required by this AD if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Los Angeles

ACO, to make those findings. To be approved, the repair method, modification deviation, or alteration deviation must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(l) Related Information

For more information about this AD, contact Eric Schrieber, Aerospace Engineer, Airframe Branch, ANM-120L, FAA, Los Angeles ACO, 3960 Paramount Boulevard, Lakewood, CA 90712-4137; phone: 562-627-5348; fax: 562-627-5210; email: Eric.Schrieber@faa.gov.

(m) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Boeing Alert Service Bulletin 717-55A0012, dated June 12, 2015.

(ii) Reserved.

(3) For Boeing service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, 3855 Lakewood Boulevard, MC D800-0019, Long Beach, CA 90846-0001; telephone: 206-544-5000, extension 2; fax: 206-766-5683; Internet: <https://www.myboeingfleet.com>.

(4) You may view this service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221.

(5) You may view this service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Renton, Washington, on April 20, 2016.

John P. Piccola, Jr.,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2016-10160 Filed 5-3-16; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-7490; Directorate Identifier 2015-NE-40-AD; Amendment 39-18500; AD 2016-09-02]

RIN 2120-AA64

Airworthiness Directives; Turbomeca S.A. Turboshaft Engines

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for certain Turbomeca S.A. Astazou XIV B and H turboshaft engines. This AD requires a one-time inspection of the front surface of the 3rd stage turbine for a groove. This AD was prompted by a report of a crack on the 3rd stage turbine wheel. We are issuing this AD to prevent cracks in the 3rd stage turbine wheel, failure of the engine, in-flight shutdown, and loss of control of the helicopter.

DATES: This AD becomes effective June 8, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of June 8, 2016.

ADDRESSES: For service information identified in this final rule, contact Turbomeca S.A., 40220 Tarnos, France; phone: 33 (0)5 59 74 40 00; fax: 33 (0)5 59 74 45 15. You may view this service information at the FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-7490.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-7490; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the mandatory continuing airworthiness information (MCAI), the regulatory evaluation, any comments received, and other information. The address for the Docket Office (phone: 800-647-5527) is Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Wego Wang, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7134; fax: 781-238-7199; email: wego.wang@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 by adding an AD that would apply to the specified products. The

NPRM was published in the **Federal Register** on February 2, 2016 (81 FR 5395). The NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

During the overhaul of an ASTAZOU XIV engine, a crack was detected on the front face of the third stage turbine wheel between two balancing lugs. The cause of the crack is probably linked to a geometric singularity, likely caused by the transformation operation aimed at introducing expansion slots between the blades during embodiment of Turbomeca mod AB 173. Although there is only one known case of this type of crack, and although it was detected, the possibility exists that additional parts have the same geometric singularity.

This condition, if not detected and corrected, may lead to failure of a turbine blade and its associated piece of rim, possibly resulting in an uncommanded in-flight shut-down and/or release of high energy debris.

You may obtain further information by examining the MCAI in the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-7490.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (81 FR 5395, February 2, 2016).

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting this AD as proposed.

Related Service Information Under 1 CFR Part 51

Turbomeca S.A. has issued Service Bulletin (SB) No. 283 72 0811, Version A, dated August 25, 2015. The SB describes procedures for inspection of the 3rd stage turbine wheel. This service information is reasonably available because the interested parties have access to it through their normal course of business or by the means identified in the **ADDRESSES** section.

Costs of Compliance

We estimate that this AD affects 9 engines installed on helicopters of U.S. registry. We also estimate that it will take about 5 hours per engine to comply with this AD. The average labor rate is \$85 per hour. Based on these figures, we estimate the cost of this AD on U.S. operators to be \$3,825.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I,

section 106, describes the authority of the FAA Administrator. "Subtitle VII: Aviation Programs," describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in "Subtitle VII, Part A, Subpart III, Section 44701: General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866,
- (2) Is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979),
- (3) Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction, and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2016-09-02 Turbomeca S.A.: Amendment 39-18500; Docket No. FAA-2015-7490; Directorate Identifier 2015-NE-40-AD.

(a) Effective Date

This AD becomes effective June 8, 2016.

(b) Affected ADs

None.

(c) Applicability

This AD applies to all Astazou XIV B and XIV H turboshaft engines with 3rd stage turbine wheel, part number (P/N) 0 265 25 700 0 or P/N 0 265 25 706 0, installed, if the engine incorporates Turbomeca modification AB-173 or AB-208.

(d) Reason

This AD was prompted by a report of a crack on the 3rd stage turbine wheel. We are issuing this AD to prevent cracks in the 3rd stage turbine wheel, failure of the engine, in-flight shutdown, and loss of control of the helicopter.

(e) Actions and Compliance

Comply with this AD within the compliance times specified, unless already done.

(1) At the next piece part exposure of the 3rd stage turbine wheel or within 1,000 engine hours after the effective date of this AD whichever comes first, perform a one-time inspection for a groove on the front surface of the 3rd stage turbine wheel. Use Accomplishment Instructions, paragraph 4.4.2, of Turbomeca S.A. Service Bulletin (SB) No. 283 72 0811, Version A, dated August 25, 2015 to perform the inspection.

(2) If the 3rd stage turbine wheel passes inspection required by paragraph (e)(1) of this AD, no further action is required.

(3) If the 3rd stage turbine wheel fails inspection required by paragraph (e)(1) of this AD, remove the part and replace with a part eligible for installation.

(f) Installation Prohibition

After the effective date of this AD, do not install any 3rd stage turbine wheel, P/N 0 265 25 700 0 or P/N 0 265 25 706 0, unless it was inspected per the Accomplishment Instructions, paragraph 4.4.2, of Turbomeca S.A. SB No. 283 72 0811, Version A, dated August 25, 2015.

(g) Alternative Methods of Compliance (AMOCs)

The Manager, Engine Certification Office, FAA, may approve AMOCs for this AD. Use the procedures found in 14 CFR 39.19 to make your request. You may email your request to: ANE-AD-AMOC@faa.gov.

(h) Related Information

(1) For more information about this AD, contact Wego Wang, Aerospace Engineer, Engine Certification Office, FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA 01803; phone: 781-238-7134; fax: 781-238-7199; email: wego.wang@faa.gov.

(2) Refer to MCAI European Aviation Safety Agency AD 2015-0223, dated November 16, 2015, for more information. You may examine the MCAI in the AD

docket on the Internet at <http://www.regulations.gov/>
#!documentDetail;D=FAA-2015-7490-0001.

(i) Material Incorporated by Reference

(1) The Director of the Federal Register approved the incorporation by reference (IBR) of the service information listed in this paragraph under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) You must use this service information as applicable to do the actions required by this AD, unless the AD specifies otherwise.

(i) Turbomeca S.A. Service Bulletin No. 283 72 0811, Version A, dated August 25, 2015.

(ii) Reserved.

(3) For Turbomeca S.A. service information identified in this AD, contact Turbomeca S.A., 40220 Tarnos, France; phone: 33 (0)5 59 74 40 00; fax: 33 (0)5 59 74 45 15.

(4) You may view this service information at FAA, Engine & Propeller Directorate, 1200 District Avenue, Burlington, MA. For information on the availability of this material at the FAA, call 781-238-7125.

(5) You may view this service information at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: <http://www.archives.gov/federal-register/cfr/ibr-locations.html>.

Issued in Burlington, Massachusetts, on April 21, 2016.

Colleen M. D'Alessandro,

*Manager, Engine & Propeller Directorate,
Airframe Certification Service.*

[FR Doc. 2016-10279 Filed 5-3-16; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2015-5811; Directorate Identifier 2014-NM-158-AD; Amendment 39-18489; AD 2016-08-13]

RIN 2120-AA64

**Airworthiness Directives; Airbus
Airplanes**

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are superseding Airworthiness Directive (AD) 2004-19-11 for certain Airbus Model A320 series airplanes. AD 2004-19-11 required modification of the inner rear spar web of the wing, cold expansion of the attachment holes of the forward pintle fitting and the actuating cylinder anchorage of the main landing gear (MLG), repetitive ultrasonic inspections for cracking of the rear spar of the wing,

and corrective action if necessary. AD 2004-19-11 also provided optional terminating action for the repetitive inspections. This new AD retains the requirements of AD 2004-19-11, and requires the previously optional terminating action. This AD was prompted by a determination that the previously optional terminating action is necessary to address the unsafe condition. We are issuing this AD to prevent fatigue cracking of the inner rear spar, which may lead to reduced structural integrity of the wing and the MLG.

DATES: This AD becomes effective June 8, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of June 8, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of November 5, 2004 (69 FR 58828, October 1, 2004).

The Director of the Federal Register approved the incorporation by reference of certain other publications listed in this AD as of June 30, 2000 (65 FR 34069, May 26, 2000).

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of February 14, 1994 (59 FR 1903, January 13, 1994).

The Director of the Federal Register approved the incorporation by reference of a certain other publication listed in this AD as of June 11, 1993 (58 FR 27923, May 12, 1993).

ADDRESSES: For service information identified in this final rule, contact Airbus, Airworthiness Office—ELIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-5811.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2015-5811; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday,

except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (telephone 800-647-5527) is Docket Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Sanjay Ralhan, Aerospace Engineer, International Branch, ANM-116, Transport Airplane Directorate, FAA, 1601 Lind Avenue SW., Renton, WA 98057-3356; telephone 425-227-1405; fax 425-227-1149.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to supersede AD 2004-19-11, Amendment 39-13805 (69 FR 58828, October 1, 2004) (“AD 2004-19-11”). AD 2004-19-11 applied to certain Airbus Model 320 series airplanes. The NPRM published in the **Federal Register** on November 27, 2015 (80 FR 74058) (“the NPRM”). The NPRM was prompted by a determination that the previously optional terminating action is necessary to address the unsafe condition. The NPRM proposed to retain the requirements of AD 2004-19-11, and requires the previously optional terminating action. We are issuing this AD to prevent fatigue cracking of the inner rear spar, which may lead to reduced structural integrity of the wing and the MLG.

The European Aviation Safety Agency (EASA), which is the Technical Agent for the Member States of the European Union, has issued EASA Airworthiness Directive 2014-0169, dated July 17, 2014, corrected July 22, 2014 (referred to after this as the Mandatory Continuing Airworthiness Information, or “the MCAI”), to correct an unsafe condition on certain Airbus Model 320 series airplanes. The MCAI states:

During centre fuselage certification full scale fatigue test, cracks were found on the inner rear spar at holes position 52 on the right hand wing due to fatigue aspects.

This condition, if not detected and corrected, could affect the structural integrity of the aeroplane.

To prevent such cracks, Airbus developed modifications, which were introduced in production and in service through several Airbus Service Bulletins (SB).

DGAC France issued * * * [an earlier AD], which was subsequently superseded by [DGAC] AD 2001-249 [which corresponds with FAA AD 2004-19-11, Amendment 39-13805 (69 FR 58828, October 1, 2004)], to require modification of the rear spar on some