

compact fluorescent lamps, or integrated LED lamps.

2. Definitions:

Measured initial input power means the input power to the lamp, measured after the lamp is stabilized and seasoned (if applicable), and expressed in watts (W).

Measured initial lumen output means the lumen output of the lamp, measured after the lamp is stabilized and seasoned (if applicable), and expressed in lumens (lm).

Power factor means the measured initial input power (watts) divided by the product of the input voltage (volts) and the input current (amps) measured at the same time as the initial input power.

3. Active Mode Test Procedures

3.1. Take measurements at full light output.

3.2. Do not use a goniophotometer.

3.3. For single base OLED and non-integrated LED lamps, position a lamp in either the base-up and base-down orientation throughout testing. Test an equal number of lamps in the sample in the base-up and base-down orientations, except that, if the manufacturer restricts the orientation, test all of the units in the sample in the manufacturer-specified orientation. For double base OLED and non-integrated LED lamps, test all units in the horizontal orientation except that, if the manufacturer restricts the orientation, test all of the units in the sample in the manufacturer-specified orientation.

3.4. Operate the lamp at the rated voltage throughout testing. For lamps with multiple rated voltages including 120 volts, operate the lamp at 120 volts. If a lamp is not rated for 120 volts, operate the lamp at the highest

rated input voltage. For non-integrated LED lamps, operate the lamp at the manufacturer-declared input voltage and current.

3.5. Operate the lamp at the maximum input power. If multiple modes occur at the same maximum input power (such as variable CCT or CRI), the manufacturer may select any of these modes for testing; however, all measurements must be taken at the same selected mode. The manufacturer must indicate in the test report which mode was selected for testing and include detail such that another laboratory could operate the lamp in the same mode.

3.6. To measure initial lumen output, input power, input voltage, and input current use the test procedures in the table in this section.

TABLE 3.1—REFERENCES TO INDUSTRY STANDARD TEST PROCEDURES

Lamp type	Referenced test procedure
General service incandescent lamps	Appendix R to subpart B of 10 CFR part 430.
Compact fluorescent lamps	Appendix W to subpart B of 10 CFR part 430.
Integrated LED lamps	Appendix BB to subpart B of 10 CFR part 430.
Other incandescent lamps that are not reflector lamps	IES LM-45-15, sections 4-6, and section 7.1.*
Other incandescent lamps that are reflector lamps	IES LM-20-13, sections 4-6, and section 8.*
Other fluorescent lamps	IES LM-9-09-DD, sections 4-6, and section 7.5.*
OLED lamps	IES LM-79-08-DD, sections 1.3 (except 1.3f), 2.0, 3.0, 5.0, 7.0, 8.0, 9.1 and 9.2.*
Non-integrated LED lamps	IES LM-79-08-DD, sections 1.3 (except 1.3f), 2.0, 3.0, 5.0, 7.0, 8.0, 9.1 and 9.2.*

* Incorporated by reference, see § 430.3.

3.7. Determine initial lamp efficacy by dividing the measured initial lumen output (lumens) by the measured initial input power (watts).

3.8. Determine power factor by dividing the measured initial input power (watts) by the product of the measured input voltage (volts) and measured input current (amps).

4. Standby Mode Test Procedure

4.1. Measure standby mode power only for lamps that are capable of standby mode operation.

4.2. Maintain lamp orientation as specified in section 3.3 of this appendix.

4.3. Connect the lamp to the manufacturer-specified wireless control network (if applicable) and configure the lamp in standby mode by sending a signal to the lamp instructing it to have zero light output. Lamp must remain connected to the network throughout testing.

4.4. Operate the lamp at the rated voltage throughout testing. For lamps with multiple rated voltages including 120 volts, operate the lamp at 120 volts. If a lamp is not rated for 120 volts, operate the lamp at the highest rated input voltage.

4.5. Stabilize the lamp prior to measurement as specified in section 5 of IEC 62301-DD (incorporated by reference; see § 430.3).

4.6. Measure the standby mode power in watts as specified in section 5 of IEC 62301-DD (incorporated by reference; see § 430.3).

[FR Doc. 2016-25180 Filed 10-19-16; 8:45 am]

BILLING CODE 6450-01-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2014-0578; Directorate Identifier 2013-SW-048-AD; Amendment 39-18684; AD 2016-21-03]

RIN 2120-AA64

Airworthiness Directives; Airbus Helicopters Deutschland GmbH (Previously Eurocopter Deutschland GmbH) (Airbus Helicopters) Helicopters

AGENCY: Federal Aviation Administration (FAA), DOT.

ACTION: Final rule.

SUMMARY: We are adopting a new airworthiness directive (AD) for Airbus Helicopters Model MBB-BK 117 C-2 helicopters with certain duplex trim actuators installed. This AD requires repetitively inspecting the lateral and longitudinal trim actuator output levers for correct torque of the nuts. This AD was prompted by a design review that indicated the attachment screws can become loose under certain circumstances. These actions are intended to prevent the loss of an attachment screw, which could result in

movement of the output lever in an axial direction, contact of a bolt connecting the control rod to an output lever with the actuator housing, and subsequent loss of helicopter control.

DATES: This AD is effective November 25, 2016.

ADDRESSES: For service information identified in this final rule, contact Airbus Helicopters, Inc., 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>. You may review the referenced service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Pkwy, Room 6N-321, Fort Worth, TX 76177.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the Docket Operations Office between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the European Aviation Safety Agency (EASA) AD, the economic evaluation, any comments received, and other information. The street address for the Docket Operations Office (phone: 800-647-5527) is U.S. Department of Transportation, Docket

Operations Office, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT: Matt Wilbanks, Aviation Safety Engineer, Regulations and Policy Group, Rotorcraft Directorate, FAA, 10101 Hillwood Pkwy., Room 6N-321, Fort Worth, TX 76177; telephone (817) 222-5110; email matt.wilbanks@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

On August 18, 2014, at 79 FR 48696, the **Federal Register** published our notice of proposed rulemaking (NPRM), which proposed to amend 14 CFR part 39 by adding an AD that would apply to Airbus Helicopters Model MBB-BK 117 C-2 helicopters with a lateral duplex trim actuator, part number (P/N) 418-00878-050 or P/N 418-00878-051, or a longitudinal duplex trim actuator, P/N 418-00878-000 or P/N 418-00878-001, installed. The NPRM proposed to require repetitively inspecting the lateral and longitudinal trim actuator output levers for correct torque of the nuts. The proposed requirements were intended to prevent a loose attachment screw, which could result in movement of the output lever in an axial direction, contact of a bolt connecting the control rod to an output lever with the actuator housing, and subsequent loss of helicopter control.

The NPRM was prompted by AD No. 2013-0182, dated August 12, 2013, issued by EASA, which is the Technical Agent for the Member States of the European Union, to correct an unsafe condition for Airbus Helicopters Model MBB-BK 117 C-2 helicopters with a lateral duplex trim actuator, P/N 418-00878-050 or P/N 418-00878-051, or with a longitudinal duplex trim actuator, P/N 418-00878-000 or P/N 418-00878-001. EASA advises that under unfavorable circumstances, a total loss of the trim actuator output lever attachment screw could lead to a restriction of the lateral and longitudinal control range. According to EASA, without the attachment screw, the output lever can move in the axial direction. This condition, if not detected, could cause the bolt that connects the control rod to the output lever to make contact with actuator housing, possibly resulting in reduced control of the helicopter.

To prevent this condition, EASA requires an initial torque check of the lateral and longitudinal trim actuator output lever attachment screws, the application of a torque marking, and repetitive inspections for correct torque

thereafter. The EASA AD's requirements are considered an interim solution, pending a terminating modification.

Since the issuance of EASA AD No. 2013-0182, Eurocopter Deutschland GmbH has changed its name to Airbus Helicopters Deutschland GmbH.

Comments

We gave the public the opportunity to participate in developing this AD, but we received no comments on the NPRM (79 FR 48696, August 18, 2014).

FAA's Determination

These helicopters have been approved by the aviation authority of Germany and are approved for operation in the United States. Pursuant to our bilateral agreement with Germany, EASA, its technical representative, has notified us of the unsafe condition described in the EASA AD. We are issuing this AD because we evaluated all information provided by EASA and determined the unsafe condition exists and is likely to exist or develop on other helicopters of these same type designs and that air safety and the public interest require adopting the AD requirements as proposed.

Interim Action

We consider this AD to be an interim action because Airbus Helicopters is currently developing a modification that will address the unsafe condition identified in this AD. Once this modification is developed, approved and available, we might consider additional rulemaking.

Related Service Information

We reviewed Eurocopter (now Airbus Helicopters) Alert Service Bulletin MBB-BK117 C-2-67A-020, Revision 0, dated June 18, 2013 (ASB), which advises of a design review that showed that a loss of the attachment screw of the trim actuator output lever could restrict the lateral and longitudinal control range. The ASB consequently calls for an initial torque check and application of torque markings of the self-locking nuts, and subsequent repetitive inspections to maintain the proper torque.

Costs of Compliance

We estimate that this AD affects 100 helicopters of U.S. Registry and that labor costs average \$85 per work-hour. Based on these estimates, we expect the following costs:

- Applying torque and torque marking to the lateral and longitudinal trim actuator output levers requires 1 work-hour for a labor cost of \$85. No

parts are needed, so the cost for the U.S. fleet totals \$8,500.

- Visually inspecting for correct torque requires 0.5 work-hour for a labor cost of about \$43. No parts are needed, so the total cost for the U.S. fleet is \$4,300 per inspection cycle.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, section 106, describes the authority of the FAA Administrator. Subtitle VII: Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701: "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on helicopters identified in this rulemaking action.

Regulatory Findings

This AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979);
- (3) Will not affect intrastate aviation in Alaska to the extent that it justifies making a regulatory distinction; and
- (4) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared an economic evaluation of the estimated costs to comply with this AD and placed it in the AD docket.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator,

the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

■ 2. The FAA amends § 39.13 by adding the following new airworthiness directive (AD):

2016-21-03 Airbus Helicopters Deutschland GmbH (Previously Eurocopter Deutschland GmbH) (Airbus Helicopters) Helicopters: Amendment 39-18684; Docket No. FAA-2014-0578; Directorate Identifier 2013-SW-048-AD.

(a) Applicability

This AD applies to Airbus Helicopters Model MBB-BK 117 C-2 helicopters with a lateral duplex trim actuator, part number (P/N) 418-00878-050 or P/N 418-00878-051, or a longitudinal duplex trim actuator, P/N 418-00878-000 or P/N 418-00878-001, installed, certificated in any category.

(b) Unsafe Condition

This AD defines the unsafe condition as loss of a trim actuator output lever attachment screw. This condition could result in movement of the output lever in an axial direction, contact of a bolt connecting the control rod to an output lever with the actuator housing, and subsequent loss of control of the helicopter.

(c) Effective Date

This AD becomes effective November 25, 2016.

(d) Compliance

You are responsible for performing each action required by this AD within the specified compliance time unless it has already been accomplished prior to that time.

(e) Required Actions

(1) Within 300 hours time-in-service (TIS), apply a torque of 31.0 inch-pounds (3.5 Nm) to the self-locking nut (nut) on each lateral and longitudinal trim actuator output lever and apply a torque marking between the nut and the screw.

(2) Thereafter at intervals not to exceed 400 hours TIS, visually inspect each nut on each lateral and longitudinal trim actuator output lever to determine whether the torque is at 31.0 inch-pounds (3.5 Nm). If the torque is not at 31.0 inch-pounds, apply a torque of 31.0 inch-pounds (3.5 Nm), remove the previous torque marking, and apply a new torque marking between the nut and the screw.

(3) Do not install a lateral duplex trim actuator, part number (P/N) 418-00878-050 or P/N 418-00878-051, or a longitudinal duplex trim actuator, P/N 418-00878-000 or P/N 418-00878-001, on any helicopter unless each nut has been inspected for proper torque in accordance with the requirements of this AD.

(f) Alternative Methods of Compliance (AMOCs)

(1) The Manager, Safety Management Group, FAA, may approve AMOCs for this AD. Send your proposal to: Matt Wilbanks, Aviation Safety Engineer, Regulations and Policy Group, Rotorcraft Directorate, FAA, 10101 Hillwood Pkwy., Room 6N-321, Fort Worth, TX 76177; telephone (817) 222-5110; email 9-ASW-FTW-AMOC-Requests@faa.gov.

(2) For operations conducted under a 14 CFR part 119 operating certificate or under 14 CFR part 91, subpart K, we suggest that you notify your principal inspector, or lacking a principal inspector, the manager of the local flight standards district office or certificate holding district office, before operating any aircraft complying with this AD through an AMOC.

(g) Additional Information

(1) Eurocopter Alert Service Bulletin MBB-BK117 C-2-67A-020, Revision 0, dated June 18, 2013, which is not incorporated by reference, contains additional information about the subject of this AD. For service information identified in this final rule, contact Airbus Helicopters, Inc., 2701 N. Forum Drive, Grand Prairie, TX 75052; telephone (972) 641-0000 or (800) 232-0323; fax (972) 641-3775; or at <http://www.airbushelicopters.com/techpub>. You may review a copy of the service information at the FAA, Office of the Regional Counsel, Southwest Region, 10101 Hillwood Parkway, Room 6N-321, Fort Worth, TX 76177.

(2) The subject of this AD is addressed in the European Aviation Agency (EASA) AD No. 2013-0182, dated August 12, 2013. You may view the EASA AD on the Internet at <http://www.regulations.gov> in Docket No. FAA-2014-0578.

(h) Subject

Joint Aircraft Service Component (JASC) Code: 6700, Rotorcraft Flight Control.

Issued in Fort Worth, Texas, on October 5, 2016.

Lance T. Gant,

Manager, Rotorcraft Directorate, Aircraft Certification Service.

[FR Doc. 2016-24860 Filed 10-19-16; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA-2016-5589; Directorate Identifier 2014-NM-252-AD; Amendment 39-18678; AD 2016-20-12]

RIN 2120-AA64

Airworthiness Directives; Airbus Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are superseding Airworthiness Directive (AD) 2012-20-07 for certain Airbus Model A318, A319, A320, and A321 series airplanes. AD 2012-20-07 required revising the Airworthiness Limitations section (ALS) of the Instructions for Continued Airworthiness (ICA) to incorporate new limitations for fuel tank systems, and revising the maintenance program to incorporate revised fuel maintenance and inspection tasks. This new AD requires revising the maintenance or inspection program to incorporate revised fuel airworthiness limitations. This AD was prompted by Airbus issuing more restrictive maintenance requirements and/or airworthiness limitations. We are issuing this AD to prevent the potential of ignition sources inside fuel tanks, which, in combination with flammable fuel vapors, could result in a fuel tank explosion and consequent loss of the airplane.

DATES: This AD is effective November 25, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in this AD as of November 25, 2016.

The Director of the Federal Register approved the incorporation by reference of a certain other publication listed in this AD as of November 21, 2012 (77 FR 63716, October 17, 2012).

The Director of the Federal Register approved the incorporation by reference of a certain other publication listed in this AD as of December 14, 2009 (74 FR 62219, November 27, 2009).

The Director of the Federal Register approved the incorporation by reference of a certain other publication listed in this AD as of August 28, 2007 (72 FR 40222, July 24, 2007).

ADDRESSES: For service information identified in this final rule, contact Airbus, Airworthiness Office—EIAS, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; email account.airworth-eas@airbus.com; Internet <http://www.airbus.com>. You may view this referenced service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue SW., Renton, WA. For information on the availability of this material at the FAA, call 425-227-1221. It is also available on the Internet at <http://www.regulations.gov> by searching for and locating Docket No. FAA-2016-5589.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov> by searching for