

accomplish those actions in the future if this AD were not adopted.

### Regulatory Impact

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

### The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

### PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

#### § 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

**Airbus Industrie:** Docket 98–NM–75–AD.

**Applicability:** Model A319 series airplanes, manufacturer's serial numbers 578 through 625 inclusive; and Model A321–100 series airplanes, manufacturer's serial numbers 385 through 620 inclusive; certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For

airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent non-extension of one or more landing gears, consequent damage to the airplane structure, and possible injury to passengers and crewmembers, accomplish the following:

(a) Within 400 flight hours after the effective date of this AD, accomplish the actions required by paragraphs (a)(1), (a)(2), (a)(3), and (a)(4) of this AD, in accordance with Airbus Industrie A319/A321 All Operator Telex (AOT) 32–15, dated July 1, 1997.

(1) Adjust the landing gear unlocked-stop screw.

(2) Replace the shear pins in the reduction gear box and the landing gear pulley assembly with new or serviceable shear pins.

(3) Inspect the cut-out valve for discrepancies. If any discrepancy to the cut-out valve is detected, accomplish the requirements of paragraphs (a)(3)(i) and (a)(3)(ii) of this AD at the time specified in the AOT.

(i) Replace the cut-out valve with a new or serviceable part within the time specified in the AOT.

(ii) After replacing the cut-out valve, perform a functional test of the normal extension and retraction of the landing gear and of the free-fall extension system. If any discrepancy is detected during the accomplishment of either of the functional tests, prior to further flight, repair in accordance with the AOT.

(4) Perform an operational test of the gear uplock and door unlock mechanical control system. If any discrepancy is detected during the accomplishment of the operational test, prior to further flight, repair in accordance with the AOT.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM–116.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM–116.

(c) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

**Note 3:** The subject of this AD is addressed in French airworthiness directive 97–177–101(B), dated August 13, 1997.

Issued in Renton, Washington, on April 15, 1998.

**Darrell M. Pederson,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 98–10476 Filed 4–20–98; 8:45 am]

BILLING CODE 4910–13–U

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 98–NM–86–AD]

RIN 2120–AA64

### Airworthiness Directives; British Aerospace (Jetstream) Model 4100 Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain British Aerospace (Jetstream) Model 4100 airplanes. This proposal would require an eddy current conductivity test to measure the conductivity of the upper splice plate of the wing, and follow-on actions, if necessary. This proposal is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by the proposed AD are intended to correct corrosion of the upper splice plate of the wing, which could result in reduced structural integrity of the airplane.

**DATES:** Comments must be received by May 21, 1998.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 98–NM–86–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from AI(R) American Support, Inc., 13850 Mclearen Road, Herndon, Virginia 20171. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

**FOR FURTHER INFORMATION CONTACT:**

Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:****Comments Invited**

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 98-NM-86-AD." The postcard will be date stamped and returned to the commenter.

**Availability of NPRMs**

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 98-NM-86-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

**Discussion**

The Civil Aviation Authority (CAA), which is the airworthiness authority for the United Kingdom, notified the FAA that an unsafe condition may exist on certain British Aerospace (Jetstream) Model 4100 airplanes. The CAA advises that it has received reports of exfoliation corrosion of the upper splice plate of the wing on certain airplanes. Investigation has revealed that the susceptibility to corrosion of the upper splice plate is related to the conductivity of the material. Because the manufacturer finds that such corrosion may be related

to a material batch problem, the corrosion is likely to be present or develop on other airplanes with an upper splice plate made from the same material. Corrosion on the upper splice plate of the wing, if not corrected, could result in reduced structural integrity of the airplane.

**Explanation of Relevant Service Information**

British Aerospace has issued Regional Aircraft Service Bulletin J41-57-019, Revision 1, dated November 26, 1997, including Appendix 1, which describes procedures for performing an eddy current conductivity test to measure the conductivity of the upper splice plate of the wing. If the conductivity of the upper splice plate of the wing is less than 35% of the International Aluminum and Copper Standards (IACS), follow-on actions are required. The CAA approved this service bulletin.

British Aerospace also has issued Regional Aircraft Service Bulletin J41-57-020, dated March 20, 1997, including Appendix 1 and Appendix 2, which describes procedures for performing repetitive detailed visual inspections, using a boroscope, to detect corrosion along the full length of the upper splice plate of the wing; repairing damage that is found to be within certain specified limits; and replacing the existing upper splice plate with a new upper splice plate, if necessary. Such replacement eliminates the need for the repetitive inspections. The CAA classified this service bulletin as mandatory and issued British airworthiness directive 005-03-97 (undated), in order to assure the continued airworthiness of these airplanes in the United Kingdom.

**FAA's Conclusions**

This airplane model is manufactured in the United Kingdom and is type certificated for operation in the United States under the provisions of § 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the CAA has kept the FAA informed of the situation described above. The FAA has examined the findings of the CAA, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

**Explanation of Requirements of Proposed Rule**

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same

type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the service bulletins described previously, except as discussed below.

**Differences Between Proposed Rule and Service Bulletin**

Operators should note that, although British Aerospace Regional Aircraft Service Bulletin J41-57-020 specifies that the manufacturer may be contacted for disposition of certain repair conditions, this proposal would require the repair of those conditions to be accomplished in accordance with a method approved by the FAA.

**Cost Impact**

The FAA estimates that 54 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 1 work hour per airplane to accomplish the proposed eddy current conductivity inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be \$3,240, or \$60 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

**Regulatory Impact**

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Safety.

**The Proposed Amendment**

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

2. Section 39.13 is amended by adding the following new airworthiness directive:

**British Aerospace Regional Aircraft [Formerly Jetstream Aircraft Limited; British Aerospace (Commercial Aircraft) Limited];** Docket 98–NM–86–AD.

**Applicability:** Jetstream Model 4100 airplanes, constructor's numbers 41004 through 41096 inclusive; certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To correct corrosion of the upper splice plate of the wing, which could result in reduced structural integrity of the airplane, accomplish the following:

(a) Within 6 months after the effective date of this AD, perform an eddy current conductivity test to measure the conductivity of the upper splice plate of the wing, in accordance with British Aerospace Regional Aircraft Service Bulletin J41–57–019, Revision 1, dated November 26, 1997, including Appendix 1. If the conductivity measurement is greater than or equal to 35.0% of the International Aluminum and Copper Standards (IACS), no further action is required by this AD.

(b) During the inspection required by paragraph (a) of this AD, if the conductivity measurement is less than 35.0% of the IACS: Prior to further flight, use a boroscope to perform a detailed visual inspection to detect corrosion along the full length of the upper splice plate of the wing, in accordance with

British Aerospace Regional Aircraft Service Bulletin J41–57–020, dated March 20, 1997, including Appendix 1 and Appendix 2. Thereafter, repeat the inspection at intervals not to exceed 1 year.

(1) During any inspection required by paragraph (b) of this AD, if any corrosion is detected that is within the allowable limits specified in the service bulletin: Accomplish the actions required by paragraphs (b)(1)(i) and (b)(1)(ii) of this AD, at the times specified in those paragraphs.

(i) Prior to further flight, repair the upper splice plate of the wing in accordance with Appendix 2 of the service bulletin. And

(ii) Within 3 years after the detection of corrosion, replace the upper splice plate of the wing with a new upper splice plate in accordance with the service bulletin. Such replacement constitutes terminating action for the requirements of this AD.

(2) During any inspection required by paragraph (b) of this AD, if any corrosion is detected that is outside the allowable limits specified in the service bulletin: Prior to further flight, repair in accordance with a method approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate.

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM–116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM–116.

**Note 2:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM–116.

(d) Special flight permits may be issued in accordance with §§ 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

**Note 3:** The subject of this AD is addressed in British airworthiness directive 005–03–97 (undated).

Issued in Renton, Washington, on April 15, 1998.

**Darrell M. Pederson,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. 98–10483 Filed 4–20–98; 8:45 am]

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**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. 98–NM–51–AD]

RIN 2120–AA64

**Airworthiness Directives; British Aerospace BAC 1–11 200 and 400 Series Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Notice of proposed rulemaking (NPRM).

**SUMMARY:** This document proposes the adoption of a new airworthiness directive (AD) that is applicable to all British Aerospace Model BAC 1–11 200 and 400 series airplanes. This proposal would require repetitive detailed visual inspections to detect cracking in the trunnion fittings located in the nose landing gear (NLG) bay of the forward fuselage; and repair, if necessary. This proposal is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by the proposed AD are intended to detect and correct cracking in the trunnion fittings of the NLG, which could lead to collapse of the NLG during takeoff and landing, and possible injury to the flight crew and passengers.

**DATES:** Comments must be received by May 21, 1998.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 98–NM–51–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9:00 a.m. and 3:00 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from British Aerospace, Service Support, Airbus Limited, P.O. Box 77, Bristol BS99 7AR, England. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington.

**FOR FURTHER INFORMATION CONTACT:** Norman B. Martenson, Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055–4056; telephone (425) 227–2110; fax (425) 227–1149.

**SUPPLEMENTARY INFORMATION:**