

wing landing gear trunnion at intervals not to exceed 6 months, until the terminating action is accomplished. Doing paragraph (b), (c), or (d) of this AD ends the repetitive inspections required by this paragraph.

Optional Terminating Action for Requirements of Paragraph (a)

(b) For airplanes listed in Groups 1, 2, and 3 in Boeing Service Bulletin 747-32-2190, Revision 4, dated October 26, 1989; Modification in accordance with Boeing Service Bulletin 747-32-2190, Revision 4, dated October 26, 1989, constitutes terminating action for the reinspection requirements of paragraph (a) of this AD.

New Requirements of This AD

Repetitive Detailed Visual Inspections and Follow-On Actions (All Airplanes)

(c) Within 180 days since the airplane's date of delivery or 180 days after the effective date of this AD, whichever occurs later, do a detailed visual inspection using a borescope to find cracking and corrosion of the aft trunnion outer cylinders of the wing landing gear. Do the inspection per Figure 2 of Boeing Alert Service Bulletin 747-32A2465, Revision 1, dated July 20, 2000. The detailed visual inspection is contained in Part 1 of the service bulletin. Thereafter, repeat the inspection at intervals not to exceed 6 months.

(1) If no corrosion or cracking is found during any inspection per paragraph (c) of this AD, before further flight, apply corrosion preventative compound, per the service bulletin. Repeat the application of corrosion preventative compound after each inspection per paragraph (c) of this AD.

(2) If any corrosion or cracking is found during any inspection per paragraph (c) of this AD, before further flight, repair per a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA; or per data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative (DER) who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the approval letter must specifically reference this AD.

Note 2: For the purposes of this AD, a detailed visual inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

Overhaul (All Airplanes)

(d) At the applicable compliance time stated in paragraph (d)(1) or (d)(2) of this AD, and thereafter at intervals not to exceed 10 years, overhaul the wing landing gear per Flag Note 2 of Figure 1 of Boeing Alert Service Bulletin 747-32A2465, Revision 1, dated July 20, 2000. If any cracking or corrosion outside the overhaul limits is

found during this overhaul, before further flight, repair per a method approved by the Manager, Seattle ACO; or per data meeting the type certification basis of the airplane approved by a Boeing Company DER who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved by the Manager, Seattle ACO, as required by this paragraph, the approval letter must specifically reference this AD. For affected airplanes, doing this overhaul ends the repetitive inspections required by paragraph (a) of this AD.

(1) For Group 1 airplanes listed in Boeing Alert Service Bulletin 747-32A2465, Revision 1, on which the wing landing gear has NOT been modified per Flag Note 1 of Figure 1 of the service bulletin: Overhaul the wing landing gear within 48 months after the effective date of this AD.

(2) For Group 1 airplanes listed in Boeing Alert Service Bulletin 747-32A2465, Revision 1, on which the wing landing gear HAS been modified per Flag Note 1 of Figure 1 of the service bulletin; OR for Groups 2 and 3 airplanes listed in Boeing Alert Service Bulletin 747-32A2465, Revision 1: Overhaul the wing landing gear within 10 years since delivery of the airplane or last overhaul, or within 180 days after the effective date of this AD, whichever comes later.

Alternative Methods of Compliance

(e)(1) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

(2) Alternative methods of compliance, approved previously in accordance with AD 90-06-18 R1, amendment 39-6706, are approved as alternative methods of compliance for paragraphs (a) and (b) of this AD.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permits

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(g) Except as provided by paragraphs (c)(2) and (d) of this AD, the actions shall be done in accordance with Boeing Service Bulletin 747-32-2190, Revision 4, dated October 26, 1989; and Boeing Alert Service Bulletin 747-32A2465, Revision 1, dated July 20, 2000; as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Boeing Commercial Airplane Group, P.O. Box 3707, Seattle, Washington 98124-2207. Copies may be inspected at the FAA,

Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Effective Date

(h) This amendment becomes effective on October 3, 2001.

Issued in Renton, Washington, on August 20, 2001.

Vi L. Lipski,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 01-21492 Filed 8-28-01; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2000-NM-373-AD; Amendment 39-12417; AD 2001-17-26]

RIN 2120-AA64

Airworthiness Directives; Raytheon Model DH.125, HS.125, BH.125, and BAe. 125 (U-125 and C-29A) Series Airplanes; Model Hawker 800, Hawker 800 (U-125A), Hawker 800XP, and Hawker 1000 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Raytheon Model DH.125, HS.125, BH.125, and BAe. 125 (U-125 and C-29A) series airplanes; Model Hawker 800, Hawker 800 (U-125A), Hawker 800XP, and Hawker 1000 airplanes, that requires an inspection for cracking or corrosion of the cylinder head lugs of the main landing gear (MLG) actuator and follow-on/corrective actions. This amendment is prompted by reports of attachment lugs cracking at the actuator cylinder head. The actions specified by this AD are intended to prevent separation of the cylinder head lugs, which could prevent the MLG from extending and result in a partial gear-up landing.

DATES: Effective October 3, 2001.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of October 3, 2001.

ADDRESSES: The service information referenced in this AD may be obtained from Raytheon Aircraft Company, Department 62, P.O. Box 85, Wichita, Kansas 67201-0085. This information may be examined at the Federal

Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Wichita, Kansas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: David Ostrodka, Aerospace Engineer, Airframe Branch, ACE-118W, FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Wichita, Kansas 67209; telephone (316) 946-4129; fax (316) 946-4407.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Raytheon Model DH.125, HS.125, BH.125, and BAe 125 (U-125 and C-29A) series airplanes, and Hawker 800, Hawker 800 (U-125A), Hawker 800XP, and Hawker 1000 airplanes was published in the *Federal Register* on June 5, 2001 (66 FR 30107). That action proposed to require an inspection to detect cracking or corrosion of the cylinder head lugs of the main landing gear (MLG) and follow-on/corrective actions.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

There are approximately 1,000 airplanes of the affected design in the worldwide fleet. The FAA estimates that 650 airplanes of U.S. registry will be affected by this AD, that it will take approximately 20 work hours per airplane to accomplish the required inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$780,000, or \$1,200 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions

actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2001-17-26 Raytheon Aircraft Company: Amendment 39-12417. Docket 2000-NM-373-AD.

Applicability: Model DH.125, HS.125, BH.125, and BAe. 125 (U-125 and C-29A) series airplanes; Model Hawker 800, Hawker 800 (U-125A), Hawker 800XP, and Hawker 1000 airplanes; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability

provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent separation of the cylinder head lugs, which could prevent the main landing gear (MLG) from extending and result in a partial gear-up landing, accomplish the following:

Inspection

(a) Perform an eddy current inspection of the actuator cylinder head lugs for cracking or corrosion per Raytheon Service Bulletin 32-3391, dated August 2000, at the time specified in paragraph (a)(1), (a)(2), (a)(3), or (a)(4) of this AD, as applicable.

(1) For actuator cylinder heads that have 3,000 or less total landings as of the effective date of this AD: Perform the eddy current inspection within 24 months after the effective date of this AD.

(2) For actuator cylinder heads that have 3,001 to 4,000 total landings as of the effective date of this AD: Perform the eddy current inspection within 6 months after the effective date of this AD.

(3) For actuator cylinder heads that have been in service for more than 7 years as of the effective date of this AD: Perform the eddy current inspection within 6 months of the effective date of this AD.

(4) For actuator cylinder heads that have 4,001 or more total landings as of the effective date of this AD: Perform the eddy current inspection within 10 landings after the effective date of this AD.

If No Cracking or Corrosion

(b) If no cracking or corrosion is found during the inspection required by paragraph (a) of this AD, before further flight, accomplish the follow-on actions (e.g., "vibro-etching" the MLG actuator data plate, painting a blue stripe on the actuator cylinder head to indicate 1/32 inch oversize bushings, replacing bushings, and applying corrosion protection to the lug bores), per Raytheon Service Bulletin 32-3391, dated August 2000.

If Any Cracking or Corrosion

(c) If any cracking or corrosion is found during the inspection required by paragraph (a) of this AD, before further flight, accomplish either of the actions specified in paragraph (c)(1) or (c)(2) of this AD, per Raytheon Service Bulletin 32-3391, dated August 2000.

(1) Replace the actuator of the MLG with a new or serviceable actuator, or

(2) Replace the actuator cylinder head with a new cylinder head.

Note 2: Raytheon Service Bulletin 32-3391, dated August 2000, references Precision Hydraulics Cylinder Maintenance Manual (CMM) 32-30-1105 as an additional source of service information.

Alternative Methods of Compliance

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Wichita Aircraft Certification Office (ACO), FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Wichita ACO.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Wichita ACO.

Special Flight Permit

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(f) The actions shall be done in accordance with Raytheon Service Bulletin 32-3391, dated August 2000. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Raytheon Aircraft Company, Department 62, P.O. Box 85, Wichita, Kansas 67201-0085. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Wichita, Kansas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Effective Date

(g) This amendment becomes effective on October 3, 2001.

Issued in Renton, Washington, on August 20, 2001.

Vi L. Lipski,

*Manager, Transport Airplane Directorate,
Aircraft Certification Service.*

[FR Doc. 01-21491 Filed 8-28-01; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001-NM-261-AD; Amendment 39-12418; AD 2001-17-27]

RIN 2120-AA64

Airworthiness Directives; Israel Aircraft Industries, Ltd., Model Astra SPX and 1125 Westwind Astra Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; request for comments.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that is applicable to certain Israel Aircraft Industries, Ltd., Model Astra SPX and 1125 Westwind Astra series airplanes. This action requires one-time detailed visual and eddy current inspections for cracking of the drag brace on the nose landing gear, and replacement of the drag brace, if necessary. This action is necessary to prevent failure of the nose landing gear. This action is intended to address the identified unsafe condition.

DATES: Effective September 13, 2001.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of September 13, 2001.

Comments for inclusion in the Rules Docket must be received on or before September 28, 2001.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2001-NM-261-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-iarcomment@faa.gov. Comments sent via fax or the Internet must contain "Docket No. 2001-NM-261-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

The service information referenced in this AD may be obtained from Galaxy Aerospace Corporation, One Galaxy Way, Fort Worth Alliance Airport, Fort Worth, Texas 76177. This information

may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2125; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: The Civil Aviation Administration of Israel (CAAI), which is the airworthiness authority for Israel, notified the FAA that an unsafe condition may exist on certain Israel Aircraft Industries, Ltd., Model Astra SPX and 1125 Westwind Astra series airplanes. The CAAI advises that fatigue cracking has been found on drag braces with part numbers 25W272003-501 through 25W272003-507 inclusive installed on the nose landing gear. This condition, if not corrected, could result in failure of the nose landing gear.

Explanation of Relevant Service Information

Israel Aircraft Industries has issued Astra Alert Service Bulletin 1125-32A-095, dated December 4, 2000. The service bulletin describes procedures for one-time detailed visual and eddy current inspections for cracking of the upper radius of the drag brace on the nose landing gear, and replacement of the drag brace, if necessary. The area to be inspected is the fillet between the trunnion axis and the body of the drag brace. Accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition. The CAAI classified this service bulletin as mandatory and issued Israeli airworthiness directive 32-00-12-14, dated July 1, 2001, in order to assure the continued airworthiness of these airplanes in Israel.

The Israeli airworthiness directive notes that the drag brace on the nose landing gear is subject to repetitive detailed visual and eddy current inspections per Chapters 5 and 32 of the Astra Maintenance Manual, including Temporary Revision 32-7, dated November 28, 2000, or a later revision. Such inspections are intended to ensure that any cracking is detected in a timely manner.

FAA's Determination

Investigation of a nose landing gear overhaul facility has led to the conclusion that a procedural and