

aft bulkhead, before further flight, repair per a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA; or per data meeting the type certification basis of the airplane approved by a Boeing Company Designated Engineering Representative (DER) who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved, as required by this paragraph, the approval must specifically reference this AD.

(2) If any discrepancy is found in any area other than that specified in paragraph (b)(1) of this AD, before further flight, do the terminating action specified in paragraph (c) of this AD.

Terminating Action

(c) Except as provided by paragraph (b)(2) of this AD, within 72 months after the effective date of this AD: Do the modification (including doing a high frequency eddy current (HFEC) inspection, oversizing the fastener holes, and installing new fasteners) as specified in and per Figure 3 of the Accomplishment Instructions of Boeing Alert Service Bulletin 747-54A2207, dated November 16, 2000. If any cracking is found during the HFEC inspection and the service bulletin specifies contacting Boeing for repair procedures, before further flight, repair per a method approved by the Manager, Seattle ACO; or per data meeting the type certification basis of the airplane approved by a Boeing Company DER who has been authorized by the Manager, Seattle ACO, to make such findings. For a repair method to be approved, as required by this paragraph, the approval must specifically reference this AD. Accomplishment of the actions specified in this paragraph ends the repetitive inspections and checks.

Alternative Methods of Compliance

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Seattle ACO. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Seattle ACO.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Seattle ACO.

Special Flight Permit

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on August 20, 2002.

Vi L. Lipski,

Manager, Transport Airplane Directorate,
Aircraft Certification Service.

[FR Doc. 02-22130 Filed 8-29-02; 8:45 am]

BILLING CODE 4910-13-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002-NM-15-AD]

RIN 2120-AA64

Airworthiness Directives; Raytheon Model DH.125, HS.125, and BH.125 Series Airplanes; Model BAe.125 Series 800A, 800A (C-29A), 800A (U-125), 800B, 1000A, and 1000B Airplanes; and Model Hawker 800, 800 (U-125A), 1000, and 800XP Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the superseding of an existing airworthiness directive (AD); applicable to certain Raytheon Model DH.125, HS.125, BH.125, and BAe.125 (U-125 and C-29A) series airplanes; and Model Hawker 800, Hawker 800 (U-125A), Hawker 800XP, and Hawker 1000 airplanes; that currently requires an inspection for cracking or corrosion of the cylinder head lugs of the main landing gear (MLG) actuator and follow-on/corrective actions. This action proposes to expand the applicability of the existing AD to add an airplane model and further clarify the applicability; and, for certain airplanes, to clarify the compliance time of the inspection requirements. The actions specified by the proposed AD are intended to prevent separation of the cylinder head lugs, which could prevent the MLG from extending and result in a partial gear-up landing.

DATES: Comments must be received by October 15, 2002.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2002-NM-15-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-nprmcomment@faa.gov. Comments sent via fax or the Internet must contain "Docket No. 2002-NM-15-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must

be formatted in Microsoft Word 97 for Windows or ASCII text.

The service information referenced in the proposed rule may be obtained from Raytheon Aircraft Company, Department 62, P.O. Box 85, Wichita, Kansas 67201-0085. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Wichita, Kansas.

FOR FURTHER INFORMATION CONTACT:

Technical Information: David Ostrodka, Aerospace Engineer, Airframe Branch, ACE-118W, FAA, Wichita Aircraft Certification Office, 1801 Airport Road, Room 100, Wichita, Kansas 67209; telephone (316) 946-4129; fax (316) 946-4407.

Other Information: Sandi Carli, Airworthiness Directive Technical Editor/Writer; telephone (425) 687-4242, fax (425) 227-1232. Questions or comments may also be sent via the Internet using the following address: sandi.carli@faa.gov. Questions or comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 for Windows or ASCII text.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the proposed AD is being requested.
- Include justification (e.g., reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by

interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2002-NM-15-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2002-NM-15-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

On January 18, 2002, the FAA issued AD 2001-17-26 R1, amendment 39-12619 (67 FR 4171, January 29, 2002), applicable to certain Raytheon Model DH.125, HS.125, BH.125, and BAe.125 (U-125 and C-29A) series airplanes; and Model Hawker 800, Hawker 800 (U-125A), Hawker 800XP, and Hawker 1000 airplanes; to require an inspection for cracking or corrosion of the cylinder head lugs of the main landing gear (MLG) actuator and follow-on/corrective actions. The requirements of that AD are intended to prevent separation of the cylinder head lugs, which could prevent the MLG from extending and result in a partial gear-up landing.

Actions Since Issuance of Previous Rule

Since the issuance of that AD, the FAA has determined that the applicability should be expanded to be consistent with the effectivity specified in Raytheon Service Bulletin 32-3391, dated August 2000 (which is referenced as the appropriate source of service information for that AD), and revised to identify model designations as published in the most recent type certificate data sheet for the affected models. In addition, we have determined that the compliance time for the inspection requirements in that AD needs to be clarified in paragraph (b) of the final rule. Therefore, we have determined that further rulemaking is necessary, and this proposed AD follows from that determination.

Explanation of Requirements of Proposed AD

Since an unsafe condition has been identified that is likely to exist or develop on other products of this same type design, the proposed AD would

supersede AD 2001-17-26 R1 to continue to require an inspection for cracking or corrosion of the cylinder head lugs of the MLG actuator, and follow-on/corrective actions. The proposed AD also would expand and clarify the applicability of the existing AD per the referenced service bulletin and type certificate data sheet, and, for certain airplanes, clarify the compliance time for the inspection requirements in paragraph (b)(3) of this AD. This proposed AD is intended to prevent separation of the cylinder head lugs, which could prevent the MLG from extending and result in a partial gear-up landing.

Explanation of Changes to AD 2001-17-26 R1

This proposed AD differs from AD 2001-17-26 R1 in that the applicability includes all of the airplane models cited in the effectivity of the referenced service bulletin. This change was necessary to include Model BAe.125 Series 800B airplanes, because that model was not specified in the applicability of AD 2001-17-26 R1. The change also clarifies model designations per the most recent type certificate data sheet.

We have further clarified the applicability of this proposed AD to specify airplane models "as listed in Raytheon Service Bulletin 32-3391, dated August 2000." This change is necessary because the effectivity of the service bulletin also specifies that accomplishment of the service bulletin is not necessary for airplanes installed with an MLG actuator having part numbers "AIR48502-5 and AIR48503-5, or DOIW00839-1 and DOIW00839-2."

We also have clarified the compliance time for the inspection requirements in paragraphs (b), (b)(3)(i), and (b)(3)(ii) of this AD for Model BAe.125 series 800B airplanes, because an actuator cylinder head could have been in service for more than 7 years and have 4,001 or more total landings as of the effective date of this AD.

Cost Impact

There are approximately 1,000 airplanes of the affected design in the worldwide fleet. We estimate that 650 airplanes of U.S. registry would be affected by this proposed AD.

The actions that are currently required by AD 2001-17-26 R1, and retained in this proposed AD, take approximately 20 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour. Based on these figures, the cost impact of the currently required actions on U.S.

operators is estimated to be \$780,000, or \$1,200 per airplane.

This proposed AD does not add any new actions or requirements, and only revises the applicability of the AD by adding an airplane model, clarifying the model designations, and clarifying the compliance time for the inspection requirements for certain airplanes. Therefore, the estimated cost impact for this proposed AD is unchanged from the existing AD.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the current or proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. However, for affected airplanes within the period under the warranty agreement, the FAA has been advised that the manufacturer has committed previously to its customers that it will bear the cost of replacement parts. The FAA also has been advised that manufacturer warranty remedies are available for labor costs associated with accomplishing the actions required by this proposed AD. Therefore, the future economic cost impact of this AD may be less than the cost impact figure indicated above. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by

contacting the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by removing amendment 39–12619 (67 FR 4171, January 29, 2002), and by adding a new airworthiness directive (AD), to read as follows:

Raytheon Aircraft Company: Docket 2002–NM–15–AD. Supersedes AD 2001–17–26 R1, Amendment 39–12619.

Applicability: Model DH.125, HS.125, and BH.125 series airplanes; Model BAe.125 series 800A, 800A (C–29A), 800A (U–125), 800B, 1000A, and 1000B airplanes; and Model Hawker 800, 800 (U–125A), 1000, and 800XP airplanes; as listed in Raytheon Service Bulletin 32–3391, dated August 2000.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (e) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent separation of the cylinder head lugs, which could prevent the main landing gear (MLG) from extending and result in a partial gear-up landing, accomplish the following:

Restatement of Certain Requirements of AD 2001–17–26 R1

Inspection

(a) For Model DH.125, HS.125, and BH.125 series airplanes; BAe.125 series 800A, 800A (C–19A), 800A (U–125A), 1000A, and 1000B airplanes; and Model Hawker 800, 800 (U–125A), 800XP, and 1000 airplanes: Perform

an eddy current inspection of the actuator cylinder head lugs for cracking or corrosion per Raytheon Service Bulletin 32–3391, dated August 2000, at the time specified in paragraph (a)(1), (a)(2), (a)(3), or (a)(4) of this AD, as applicable.

(1) For actuator cylinder heads that have 3,000 or less total landings as of October 3, 2001 (the effective date of AD 2001–17–26 R1, amendment 39–12619): Perform the eddy current inspection within 24 months after October 3, 2001.

(2) For actuator cylinder heads that have 3,001 to 4,000 total landings as of October 3, 2001: Perform the eddy current inspection within 6 months after October 3, 2001.

(3) For actuator cylinder heads that have been in service for more than 7 years as of October 3, 2001: Perform the eddy current inspection within 6 months after October 3, 2001.

(4) For actuator cylinder heads that have 4,001 or more total landings as of October 3, 2001: Perform the eddy current inspection within 10 landings after October 3, 2001.

New Requirements of This AD

(b) For Model BAe.125 series 800B airplanes: Perform an eddy current inspection of the actuator cylinder head lugs for cracking or corrosion per Raytheon Service Bulletin 32–3391, dated August 2000, at the time specified in paragraph (b)(1), (b)(2), or (b)(3) of this AD, as applicable.

(1) For actuator cylinder heads that have 3,000 or less total landings as of the effective date of this AD: Perform the eddy current inspection within 24 months after the effective date of this AD.

(2) For actuator cylinder heads that have 3,001 to 4,000 total landings as of the effective date of this AD: Perform the eddy current inspection within 6 months after the effective date of this AD.

(3) For actuator cylinder heads that have been in service for more than 7 years or that have 4,001 or more total landings as of the effective date of this AD: Perform the eddy current inspection at the earlier of the times specified in paragraph (b)(3)(i) or (b)(3)(ii) of this AD:

(i) Within 6 months after the effective date of this AD; or

(ii) Within 10 landings after the effective date of this AD.

If No Cracking or Corrosion

(c) If no cracking or corrosion is found during the inspection required by paragraph (a) or (b) of this AD, before further flight, accomplish follow-on actions (e.g., “vibro-etching” the MLG actuator data plate, painting a blue stripe on the actuator cylinder head to indicate 1/32-inch oversize bushings, replacing bushings, and applying corrosion protection to the lug bores), per Raytheon Service Bulletin 32–3391, dated August 2000.

If Any Cracking or Corrosion

(d) If any cracking or corrosion is found during the inspection required by paragraph (a) or (b) of this AD, before further flight, accomplish either of the actions specified in paragraph (d)(1) or (d)(2) of this AD, per Raytheon Service Bulletin 32–3391, dated August 2000:

- (1) Replace the actuator of the MLG with a new or serviceable actuator; or
- (2) Replace the actuator cylinder head with a new cylinder head.

Note 2: Raytheon Service Bulletin 32–3391, dated August 2000, references Precision Hydraulics Component Maintenance Manual 32–30–1105 as an additional source of service information.

Alternative Methods of Compliance

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Wichita Aircraft Certification Office (ACO), FAA. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, Wichita ACO.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Wichita ACO.

Special Flight Permit

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Issued in Renton, Washington, on August 26, 2002.

Vi L. Lipski,

*Manager, Transport Airplane Directorate,
Aircraft Certification Service.*

[FR Doc. 02–22178 Filed 8–29–02; 8:45 am]

BILLING CODE 4910–13–P

SOCIAL SECURITY ADMINISTRATION

20 CFR Part 408

RIN 0960–AF61

Special Benefits for Certain World War II Veterans

AGENCY: Social Security Administration (SSA).

ACTION: Notice of proposed rulemaking.

SUMMARY: We propose to add to our regulations a new part 408 that would set forth our rules applicable to claims for special veterans benefits (SVB) under title VIII of the Social Security Act (the Act). The title VIII program was effective in May 2000 and provides monthly benefits to certain World War II (WWII) veterans who were previously eligible for supplemental security income (SSI) payments under title XVI of the Act and reside outside the United States. These proposed rules include five new subparts that would describe: what the new part is about, how we determine whether you qualify for and are entitled to SVB, how you file for SVB, how we evaluate evidence under