

pump wiring to provide protection against chafing of the fuel pump cables. This condition, if not corrected, could generate short circuits leading to fuel pump failure and arcing. These could become a potential ignition source inside the fuel tank which, in combination with flammable fuel vapours (if present), could result in a fuel tank explosion and consequent loss of the aeroplane.

To address this unsafe condition, EASA [European Aviation Safety Agency] issued AD 2007–0066 that required this modification in accordance with Airbus Service Bulletin (SB) A300–24–0103 Revision 01. Airbus subsequently introduced an additional modification of the electrical wiring of the outer fuel pump and the landing lights of the left (LH) and the right (RH) side in Revision 02 of the SB A300–24–0103, leading to the issuance of EASA AD 2008–0188 which superseded EASA AD 2007–0066 and required the additional work.

More recently, Airbus introduced some additional protection to routes 1P and 2P harnesses in zone 571 and 671 of the aeroplane.

For the reason described above, this new AD retains the requirements of EASA AD 2008–0188, which is superseded, and requires the additional work as specified in Revision 03 of Airbus SB A300–24–0103.

The additional modification will provide additional protection from chafing and will prevent intermittent operation of the fuel pump and landing lights, as well as failure of the power supply. The modification of the wiring of the outer fuel pump and the landing light on the LH side route 1P harness and RH side route 2P harness includes additional mechanical protection that includes procedures for installing new splicing on the wires, a new cable type, shrink sleeve installation on the new wiring, and an additional braided conduit sleeve (Halar), as applicable, for the fuel pumps and the landing lights.

#### Restatement of Requirements of AD 2007–18–02, With Revised Service Information

(f) Within 31 months after October 2, 2007 (the effective date of AD 2007–18–02), unless already done, modify the inner and outer fuel pump wiring, route 1P and 2P harnesses in the LH (left-hand) wing and in the RH (right-hand) wing, in accordance with the Accomplishment Instructions of Airbus Service Bulletin A300–24–0103, Revision 01, dated January 11, 2007; or Airbus Mandatory Service Bulletin A300–24–0103, Revision 03, dated February 18, 2009. After the effective date of this AD, use only Airbus Mandatory Service Bulletin A300–24–0103, Revision 03, dated February 18, 2009. Actions done before October 2, 2007, in accordance with Airbus Service Bulletin A300–24–0103, dated March 15, 2006, for airplanes under configuration 1 as defined in Airbus Service Bulletin A300–24–0103, Revision 01, dated January 11, 2007; Revision 02, dated April 4, 2008; or Revision 03, dated February 18, 2009; are acceptable for compliance with the requirements of this paragraph.

#### New Requirements of This AD

##### Actions and Compliance

(g) Unless already done, within 12 months after the effective date of this AD, modify the wiring of the outer fuel pump and the landing light on the LH side route 1P harness and RH side route 2P harness in accordance with the Accomplishment Instructions of Airbus Mandatory Service Bulletin A300–24–0103, Revision 03, dated February 18, 2009.

##### FAA AD Differences

**Note 1:** This AD differs from the MCAI and/or service information as follows: No differences.

##### Other FAA AD Provisions

(h) The following provisions also apply to this AD:

(1) *Alternative Methods of Compliance (AMOCs):* The Manager, International Branch, ANM–116, Transport Airplane Directorate, FAA, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. Send information to ATTN: Dan Rodina, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; telephone (425) 227–2125; fax (425) 227–1149. Before using any approved AMOC on any airplane to which the AMOC applies, notify your appropriate principal inspector (PI) in the FAA Flight Standards District Office (FSDO), or lacking a PI, your local FSDO. AMOCs approved previously in accordance with AD 2007–18–02, are approved as AMOCs for the corresponding provisions of this AD.

(2) *Airworthy Product:* For any requirement in this AD to obtain corrective actions from a manufacturer or other source, use these actions if they are FAA-approved. Corrective actions are considered FAA-approved if they are approved by the State of Design Authority (or their delegated agent). You are required to assure the product is airworthy before it is returned to service.

(3) *Reporting Requirements:* For any reporting requirement in this AD, under the provisions of the Paperwork Reduction Act, the Office of Management and Budget (OMB) has approved the information collection requirements and has assigned OMB Control Number 2120–0056.

##### Related Information

(i) Refer to MCAI EASA Airworthiness Directive 2009–0157, dated July 17, 2009; Airbus Service Bulletin A300–24–0103, Revision 01, dated January 11, 2007; and Airbus Mandatory Service Bulletin A300–24–0103, Revision 03, dated February 18, 2009; for related information.

##### Material Incorporated by Reference

(j) You must use Airbus Mandatory Service Bulletin A300–24–0103, Revision 03, dated February 18, 2009, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Airbus SAS—EAW (Airworthiness Office), 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France; telephone +33 5 61 93 36 96; fax +33 5 61 93 44 51; e-mail: [account.airworth-eas@airbus.com](mailto:account.airworth-eas@airbus.com); Internet <http://www.airbus.com>.

(3) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425–227–1221 or 425–227–1152.

(4) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on November 23, 2009.

**Ali Bahrami,**

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. E9–28797 Filed 12–3–09; 8:45 am]

**BILLING CODE 4910–13–P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA–2009–0658; Directorate Identifier 2009–NM–058–AD; Amendment 39–16115; AD 2009–24–21]

**RIN 2120–AA64**

**Airworthiness Directives; McDonnell Douglas Model DC–9–14, DC–9–15, and DC–9–15F Airplanes; and McDonnell Douglas Model DC–9–20, DC–9–30, DC–9–40, and DC–9–50 Series Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** The FAA is superseding an existing airworthiness directive (AD) that applies to all McDonnell Douglas Model DC–9–14, DC–9–15, and DC–9–15F airplanes; and McDonnell Douglas Model DC–9–20, DC–9–30, DC–9–40, and DC–9–50 series airplanes. That AD currently requires repetitive inspections for cracks of the main landing gear (MLG) shock strut cylinder, and related investigative and corrective actions if necessary. This AD adds more work on airplanes that have main landing gear shock struts with certain identified part numbers. This AD results from two reports of a collapsed MLG and a report of cracks in two MLG cylinders. We are

issuing this AD to detect and correct fatigue cracks in the shock strut cylinder of the MLG, which could result in a collapsed MLG during takeoff or landing, and possible reduced structural integrity of the airplane.

**DATES:** This AD becomes effective January 8, 2010.

The Director of the Federal Register approved the incorporation by reference of a certain publication listed in the AD as of January 8, 2010.

**ADDRESSES:** For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, 3855 Lakewood Boulevard, MC D800-0019, Long Beach, California 90846-0001; telephone 206-544-5000, extension 2; fax 206-766-5683; e-mail [dse.boecom@boeing.com](mailto:dse.boecom@boeing.com); Internet <https://www.myboeingfleet.com>.

#### Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory

evaluation, any comments received, and other information. The address for the Docket Office (telephone 800-647-5527) is the Document Management Facility, U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

#### FOR FURTHER INFORMATION CONTACT:

Wahib Mina, ANM-120L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5324; fax (562) 627-5210.

#### SUPPLEMENTARY INFORMATION:

##### Discussion

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that supersedes AD 2005-19-08, amendment 39-14273 (70 FR 54616, September 16, 2005). The existing AD applies to all McDonnell Douglas Model DC-9-14, DC-9-15, and DC-9-15F airplanes; and Model DC-9-20, DC-9-30, DC-9-40, and DC-9-50 series airplanes. That NPRM was published in the **Federal Register** on July 30, 2009 (74 FR 37963).

That NPRM proposed to continue to require repetitive inspections for cracks of the main landing gear (MLG) shock strut cylinder, and related investigative and corrective actions if necessary. That NPRM also proposed to require more work on airplanes that have main landing gear shock struts with certain identified part numbers.

#### Comments

We provided the public the opportunity to participate in the development of this AD. No comments have been received on the NPRM or on the determination of the cost to the public.

#### Conclusion

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed.

#### Costs of Compliance

There are about 644 airplanes of the affected design in the worldwide fleet. The following table provides the estimated costs for U.S. operators to comply with this AD.

ESTIMATED COSTS

Action	Work hours	Average labor rate per hour	Parts	Cost per airplane	Number of U.S.-registered airplanes	Fleet cost
Inspection .....	4 to 6 .....	\$80	None .....	\$320 to \$480 per inspection cycle.	426	\$136,320 to \$204,480 per inspection cycle.

#### Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

#### Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket.

See the **ADDRESSES** section for a location to examine the regulatory evaluation.

#### List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

#### Adoption of the Amendment

■ Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

#### PART 39—AIRWORTHINESS DIRECTIVES

■ 1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

##### § 39.13 [Amended]

■ 2. The Federal Aviation Administration (FAA) amends § 39.13 by removing amendment 39-14273 (70 FR 54616, September 16, 2005) and by

adding the following new airworthiness directive (AD):

**2009–24–21 McDonnell Douglas:**

Amendment 39–16115. Docket No. FAA–2009–0658; Directorate Identifier 2009–NM–058–AD.

**Effective Date**

(a) This AD becomes effective January 8, 2010.

**Affected ADs**

(b) This AD supersedes AD 2005–19–08.

**Applicability**

(c) This AD applies to all McDonnell Douglas Model DC–9–14, DC–9–15, and DC–9–15F airplanes; Model DC–9–21 airplanes; Model DC–9–31, DC–9–32, DC–9–32 (VC–9C), DC–9–32F, DC–9–33F, DC–9–34, DC–9–34F, and DC–9–32F (C–9A, C–9B) airplanes; Model DC–9–41 airplanes; and Model DC–9–51 airplanes; certificated in any category.

**Subject**

(d) Air Transport Association (ATA) of America Code 32: Landing gear.

**Unsafe Condition**

(e) This AD results from two reports of a collapsed main landing gear (MLG) and a report of cracks in two MLG cylinders. We

are issuing this AD to detect and correct fatigue cracks in the shock strut cylinder of the MLG, which could result in a collapsed MLG during takeoff or landing, and possible reduced structural integrity of the airplane.

**Compliance**

(f) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

**Restatement of Requirements of AD 2005–19–08, With Revised Service Information**

**Records Review**

(g) Except as required by paragraph (m) of this AD, before the applicable compliance time specified in paragraph (h) or Table 1 of this AD, as applicable, do the applicable actions in paragraphs (g)(1) and (g)(2) of this AD.

(1) For all airplane groups: Review the airplane maintenance records of the MLG to determine its service history and the number of landings on the MLG shock strut cylinder.

(2) For Group 3 airplanes identified in the service bulletin: Review the maintenance records to determine if the MLG cylinder on each Group 3 airplane has always been on a Group 3 airplane, and do the actions in paragraph (k) of this AD.

**Inspection**

(h) Inspect the MLG shock strut cylinders for cracks using the Option 1 or Option 2 non-destructive testing inspection described in Boeing Alert Service Bulletin DC9–32A350, Revision 1, dated August 3, 2005; or Revision 2, dated March 20, 2009; except as required by paragraph (m) of this AD. Inspect in accordance with the Accomplishment Instructions of Boeing Alert Service Bulletin DC9–32A350, Revision 1, dated August 3, 2005; or Revision 2, dated March 20, 2009; except as required by paragraph (m) of this AD. After the effective date of this AD, use only Boeing Alert Service Bulletin DC9–32A350, Revision 2, dated March 20, 2009. Do the detailed inspection before the accumulation of 60,000 total landings on the MLG, or at the applicable grace period specified in Table 1 of this AD, whichever occurs later, except as required by paragraph (m) of this AD, and except as provided by paragraph (k) of this AD. If the review of maintenance records is not sufficient to conclusively determine the service history and number of landings on the MLG shock strut cylinder, perform the initial inspection at the applicable grace period specified in Table 1 of this AD.

TABLE 1—THRESHOLD AND REPETITIVE INTERVAL

Airplanes identified in the Service Bulletin as Group	Threshold	Repetitive interval
1 .....	Within 18 months or 650 landings after October 21, 2005 (the effective date of AD 2005–19–08), whichever occurs first.	Intervals not to exceed 650 landings.
2 .....	Within 18 months or 500 landings after October 21, 2005, whichever occurs first.	Intervals not to exceed 500 landings.
3, except as provided by paragraph (k) of this AD.	Within 18 months or 2,500 landings after October 21, 2005, whichever occurs first.	Intervals not to exceed 2,500 landings.
4 .....	Within 18 months or 2,100 landings after October 21, 2005, whichever occurs first.	Intervals not to exceed 2,100 landings.

**No Indication of Cracking Is Found**

(i) If no indication of cracking is found during the inspection required by paragraph (h) of this AD, repeat the inspection in accordance with Boeing Alert Service Bulletin DC9–32A350, Revision 1, dated August 3, 2005; or Boeing Alert Service Bulletin DC9–32A350, Revision 2, dated March 20, 2009; at the applicable interval specified in Table 1 of this AD, except as required by paragraph (m) of this AD. After the effective date of this AD, use only Boeing Alert Service Bulletin DC9–32A350, Revision 2, dated March 20, 2009.

**Related Investigative and Corrective Actions**

(j) If any indication of cracking is found during any inspection required by paragraph (h) or (i) of this AD: Before further flight, confirm the indication of cracking by doing all applicable related investigative actions and doing the applicable corrective actions in accordance with Boeing Alert Service Bulletin DC9–32A350, Revision 1, dated August 3, 2005; or Revision 2, dated March 20, 2009; except as required by paragraph (m) of this AD. After the effective date of this AD, use only Boeing Alert

Service Bulletin DC9–32A350, Revision 2, dated March 20, 2009. Repeat the inspection at the applicable threshold and interval specified in paragraph (h) of this AD.

**MLG Cylinder Previously Installed on Group 4 Airplanes**

(k) For MLG cylinders on Group 3 airplanes as identified in Boeing Alert Service Bulletin DC9–32A350, Revision 1, dated August 3, 2005; or Revision 2, dated March 20, 2009: If the MLG cylinder was previously installed on a Group 4 airplane, as identified in Boeing Alert Service Bulletin DC9–32A350, Revision 1, dated August 3, 2005; or Revision 2, dated March 20, 2009; or if the service history and number of landings cannot be determined, the MLG cylinder must be inspected at the grace period and repetitive interval that applies to Group 4 airplanes, as specified in Table 1 of this AD, except as required by paragraph (m) of this AD.

**Actions Accomplished in Accordance With Original Issue of Service Bulletin**

(l) For airplanes with shock struts that have part numbers other than 5924400–505 and

5924400–506: Actions done before the effective date of this AD in accordance with Boeing Alert Service Bulletin DC9–32A350, dated December 3, 2004, are acceptable for compliance with the corresponding actions required by paragraphs (h), (i), (j), and (k) of this by this AD.

**New Requirements of This AD**

(m) For airplanes with shock struts that have part numbers 5924400–505 and 5924400–506: Do the actions required by paragraphs (g), (h), (i), (j), and (k), as applicable, in accordance with Boeing Alert Service Bulletin DC9–32A350, Revision 2, dated March 20, 2009. Do the actions at the time specified in those paragraphs, except where Table 1 of this AD specifies a compliance time after October 21, 2005, the compliance time for these airplanes is within the specified compliance time after the effective date of this AD.

**Alternative Methods of Compliance (AMOCs)**

(n)(1) The Manager, Los Angeles Aircraft Certification Office, FAA, has the authority to approve AMOCs for this AD, if requested

using the procedures found in 14 CFR 39.19. Send information to ATTN: Wahib Mina, ANM-120L, FAA, Los Angeles Aircraft Certification Office, 3960 Paramount Boulevard, Lakewood, California 90712-4137; telephone (562) 627-5324; fax (562) 627-5210.

(2) To request a different method of compliance or a different compliance time for this AD, follow the procedures in 14 CFR 39.19. Before using any approved AMOC on any airplane to which the AMOC applies, notify your principal maintenance inspector (PMI) or principal avionics inspector (PAI), as appropriate, or lacking a principal inspector, your local Flight Standards District Office. The AMOC approval letter must specifically reference this AD.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by an Authorized Representative for the Boeing Commercial Airplanes Delegation Option Authorization Organization who has been authorized by the Manager, Los Angeles ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

#### Material Incorporated by Reference

(o) You must use Boeing Alert Service Bulletin DC9-32A350, Revision 2, dated March 20, 2009, as applicable, to do the actions required by this AD, unless the AD specifies otherwise.

(1) The Director of the Federal Register approved the incorporation by reference of this service information under 5 U.S.C. 552(a) and 1 CFR part 51.

(2) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, 3855 Lakewood Boulevard, MC D800-0019, Long Beach, California 90846-0001; telephone 206-544-5000, extension 2; fax 206-766-5683; e-mail [dse.boecom@boeing.com](mailto:dse.boecom@boeing.com); Internet <https://www.myboeingfleet.com>.

(3) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425-227-1221 or 425-227-1152.

(4) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202-741-6030, or go to: [http://www.archives.gov/federal\\_register/code\\_of\\_federal\\_regulations/ibr\\_locations.html](http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html).

Issued in Renton, Washington, on November 19, 2009.

**Stephen P. Boyd,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*

[FR Doc. E9-28564 Filed 12-3-09; 8:45 am]

**BILLING CODE 4910-13-P**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. FAA-2009-0797; Directorate Identifier 2009-CE-032-AD; Amendment 39-16118; AD 2009-25-01]

**RIN 2120-AA64**

**Airworthiness Directives; Hawker Beechcraft Corporation Models 58, 58A, 58P, 58PA, 58TC, 58TCA, 95-B55, 95-B55A, A36, A36TC, B36TC, E55, E55A, F33A, and V35B Airplanes**

**AGENCY:** Federal Aviation Administration (FAA), Department of Transportation (DOT).

**ACTION:** Final rule.

**SUMMARY:** We are adopting a new airworthiness directive (AD) to supersede AD 91-18-19, which applies to certain Hawker Beechcraft Corporation (Hawker) (Type Certificate Numbers 3A15, 3A16, and A23CE formerly held by Raytheon Aircraft Company; formerly held by Beech Aircraft Corporation) Models 58, 58A, 58P, 58PA, 58TC, 58TCA, 95-B55, 95-B55A, A36, A36TC, B36TC, E55, E55A, F33A, and V35B airplanes. AD 91-18-19 currently requires you to do a one-time inspection of the pilot and copilot shoulder harnesses for an incorrect washer and replace any incorrect washer with the correct washer. Since we issued AD 91-18-19, we have found that the applicability of AD 91-18-19 was incorrectly stated when the Model A36TC airplane was omitted from the Applicability section. Consequently, this AD would retain the actions and the serial number (SN) applicability of AD 91-18-19 and realign the SN applicability for Models A36TC and B36TC airplanes. We are issuing this AD to detect and correct an incorrect washer installed in the pilot and copilot shoulder harnesses. This incorrect part could result in a malfunctioning shoulder harness. Such a malfunction could lead to occupant injury.

**DATES:** This AD becomes effective on January 8, 2010.

As of October 21, 1991 (56 FR 42224, August 27, 1991), the Director of the Federal Register approved the incorporation by reference of Beechcraft Mandatory Service Bulletin No. 2394, dated December 1990, listed in this AD.

**ADDRESSES:** For service information identified in this AD, contact Hawker Beechcraft Corporation, P.O. Box 85, Wichita, Kansas 67201-0085; telephone: (800) 429-5372 or (316) 676-3140;

Internet: <http://pubs.hawkerbeechcraft.com>.

To view the AD docket, go to U.S. Department of Transportation, Docket Operations, M-30, West Building Ground Floor, Room W12-140, 1200 New Jersey Avenue, SE., Washington, DC 20590, or on the Internet at <http://www.regulations.gov>. The docket number is FAA-2009-0797; Directorate Identifier 2009-CE-032-AD.

#### FOR FURTHER INFORMATION CONTACT:

Steve Potter, Aerospace Engineer, 1801 Airport Road, Room 100, Wichita, Kansas 67209; telephone: (316) 946-4124; fax: (316) 946-4107.

#### SUPPLEMENTARY INFORMATION:

#### Discussion

On August 20, 2009, we issued a proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an AD that would apply to certain Hawker (Type Certificate Numbers 3A15, 3A16, and A23CE formerly held by Raytheon Aircraft Company; formerly held by Beech Aircraft Corporation) Models 58, 58A, 58P, 58PA, 58TC, 58TCA, 95-B55, 95-B55A, A36, A36TC, B36TC, E55, E55A, F33A, and V35B airplanes. This proposal was published in the **Federal Register** as a notice of proposed rulemaking (NPRM) on August 28, 2009 (74 FR 44311). The NPRM proposed to supersede AD 91-18-19 (56 FR 42224, August 27, 1991) with a new AD that would retain the actions and the SN applicability of AD 91-18-19 and realign the SN applicability for Models A36TC and B36TC airplanes.

#### Comments

We provided the public the opportunity to participate in developing this AD. We received no comments on the proposal or on the determination of the cost to the public.

#### Conclusion

We have carefully reviewed the available data and determined that air safety and the public interest require adopting the AD as proposed except for minor editorial corrections. We have determined that these minor corrections:

- Are consistent with the intent that was proposed in the NPRM for correcting the unsafe condition; and
- Do not add any additional burden upon the public than was already proposed in the NPRM.

#### Costs of Compliance

We estimate that this AD affects 4,792 airplanes in the U.S. registry.

We estimate the following costs to do the inspection: